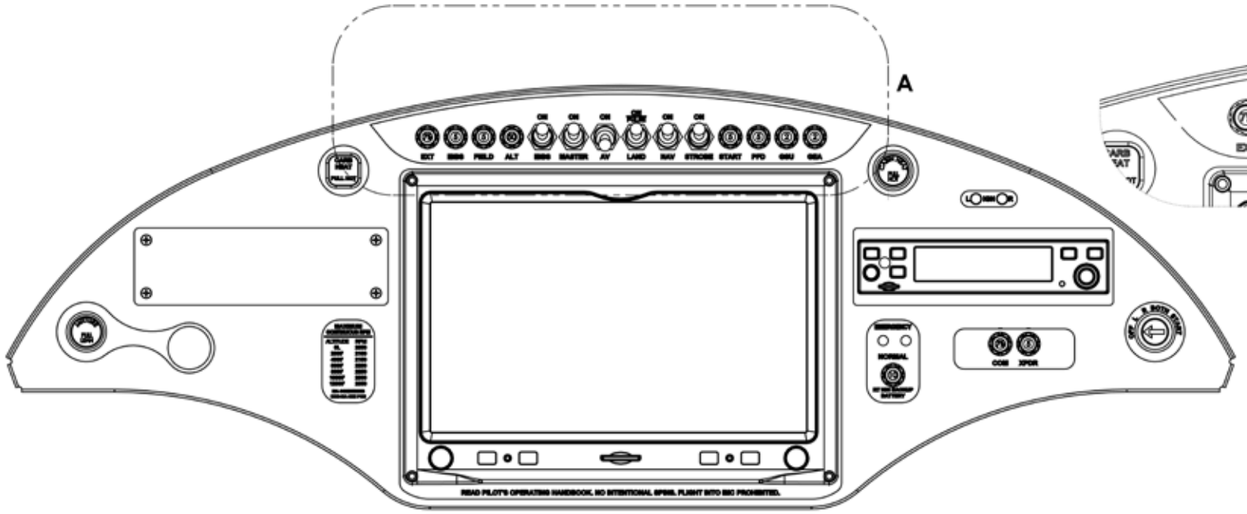
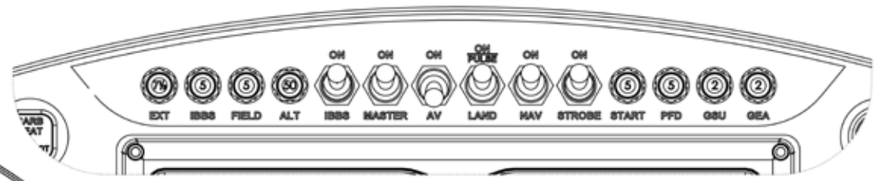
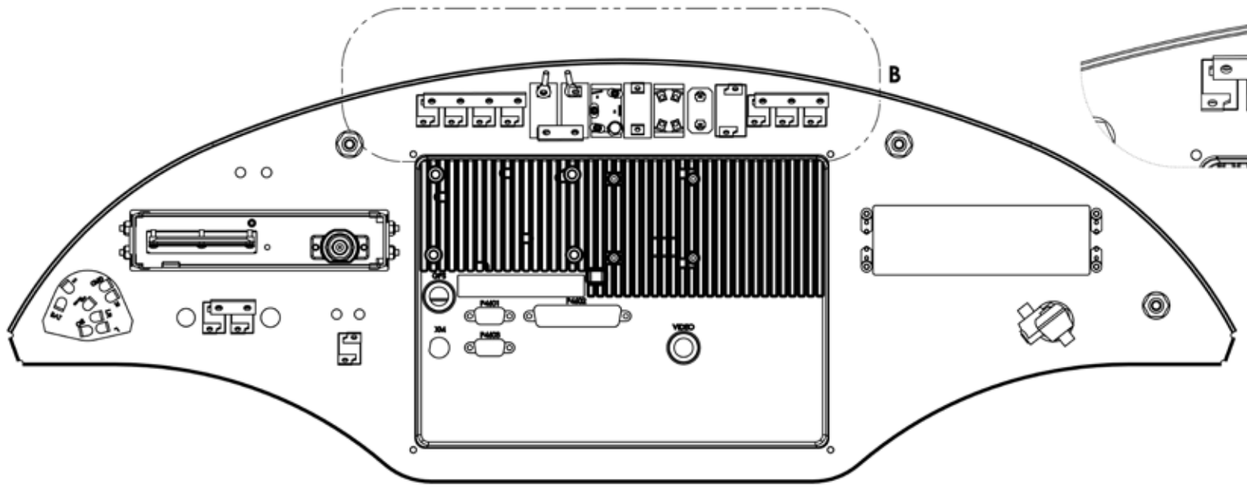
REV	CHANGE	DATE	NAME
1.06	FIRST RELEASED VERSION	05/09/2016	MST
1.07	MOVED ADAHRS TO BEHIND PANEL	06/22/2016	MST
1.08	REVISE FIGURE ON PAGE 23	09/02/2016	MST
1.09	REVISE FIGURE ON PAGE 30	09/07/2016	MST



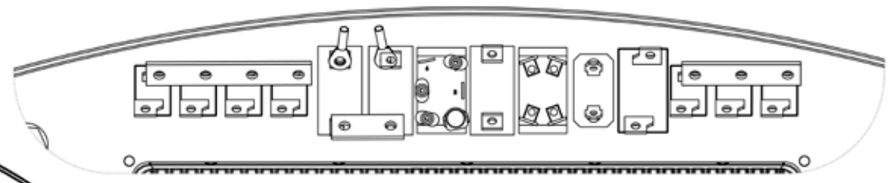
VIEW LOOKING FWD



DETAIL A
FRONT SIDE OF PANEL

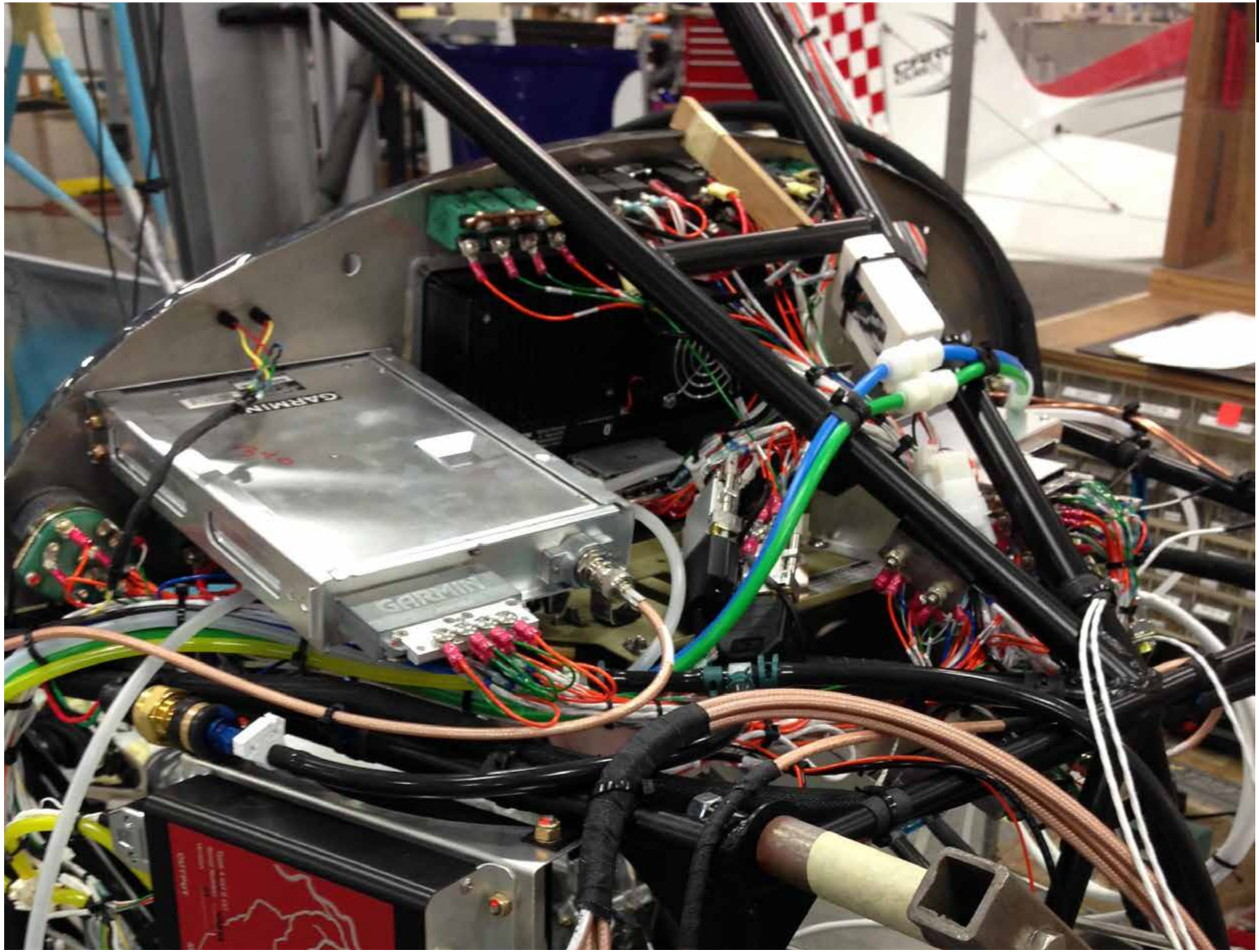


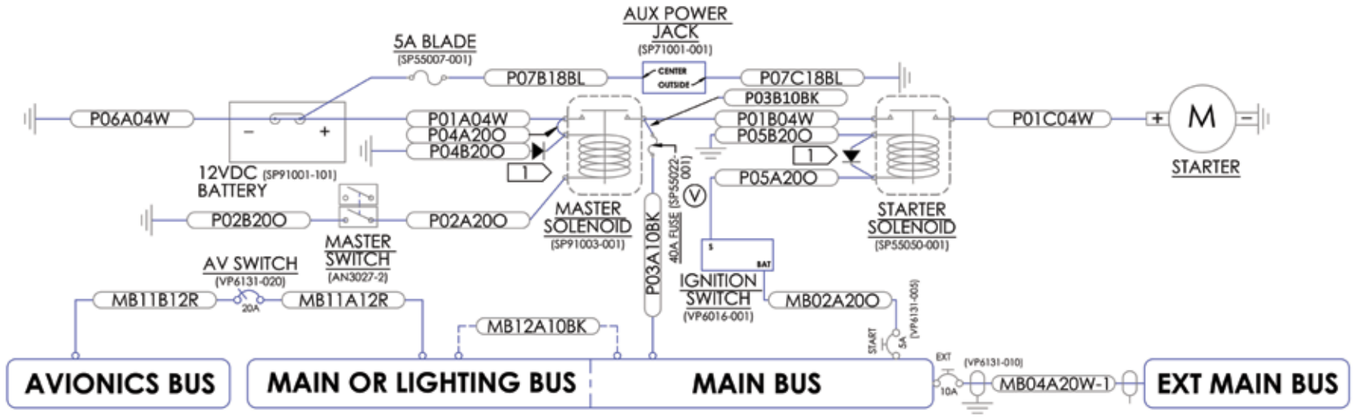
VIEW LOOKING AFT



DETAIL B
BACK SIDE OF PANEL

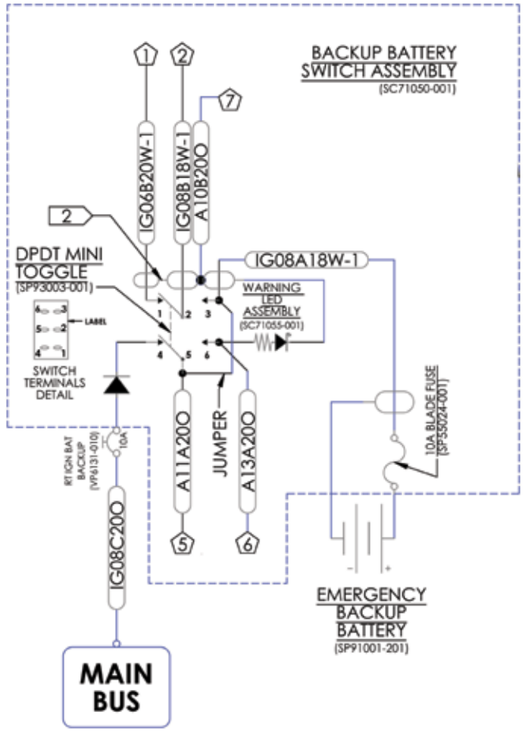
G3X INSTRUMENT PANEL ASSEMBLY



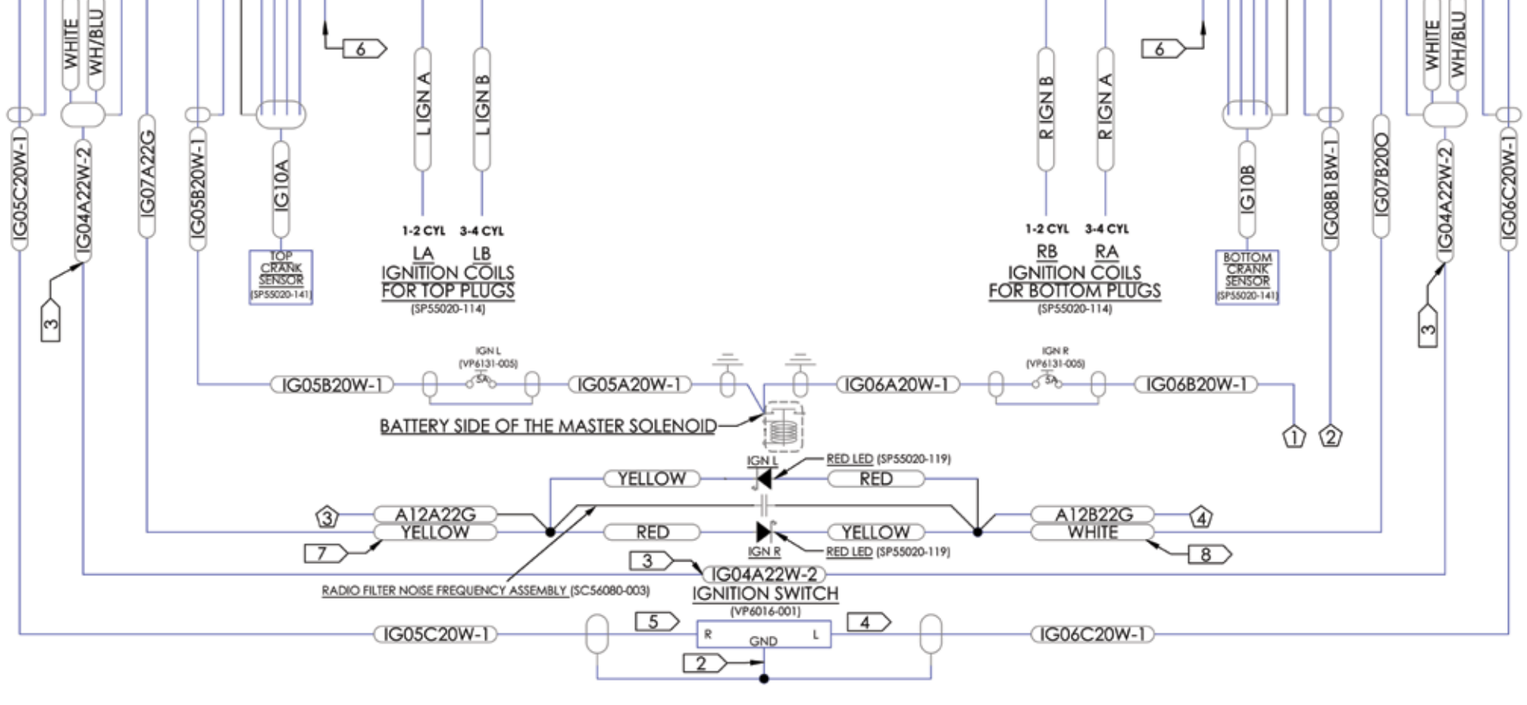


POWER DISTRIBUTION AND START CIRCUIT

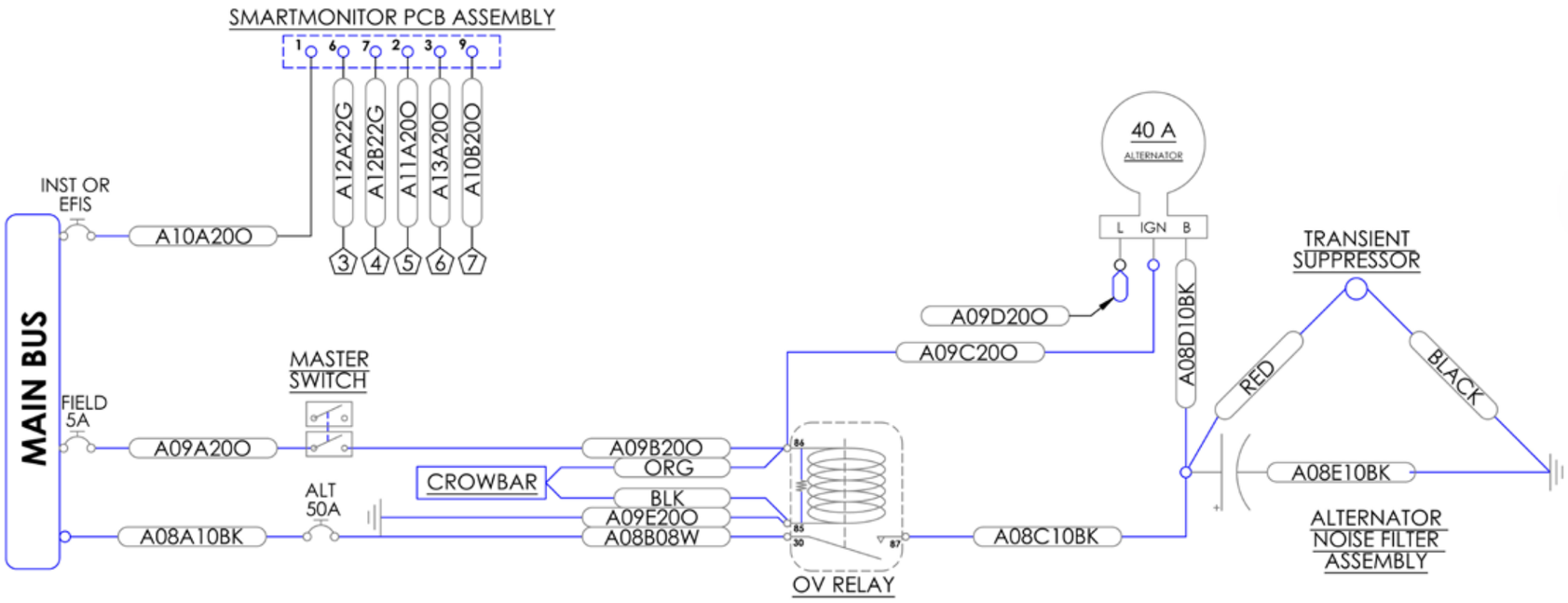
1. SURGE SUPPRESSOR DIODE.
2. USE 20 GAUGE WIRE TO JUMPER SHIELDS TO RESPECTIVE CONNECTIONS.
3. VENDOR SUPPLIED NON-MARKED WIRES.
4. RIGHT IGNITION MODULE CONNECTED TO LEFT KEYSWITCH TERMINAL.
5. LEFT IGNITION MODULE CONNECTED TO RIGHT KEYSWITCH TERMINAL.
6. OPTIONAL ELECTRONIC TACHOMETER CONNECTION POINT. MAY REQUIRE ISOLATOR. REFERENCE INSTRUMENT PANEL SCHEMATIC.
7. IG07A22G TO LEFT IGNITION
8. IG07B22G TO RIGHT IGNITION
9. IGNITION POWER WIRES MUST BE REDUNDANT, NOT SHARE ANY HARDWARE. WITH EACH OTHER. MAY SHARE HARDWARE WITH OTHER WIRES.
10. IF MAIN LIGHTING BUS NOT INSTALLED, USE MAIN BUS.
11. REFERENCE WIRE CODE LOGIC



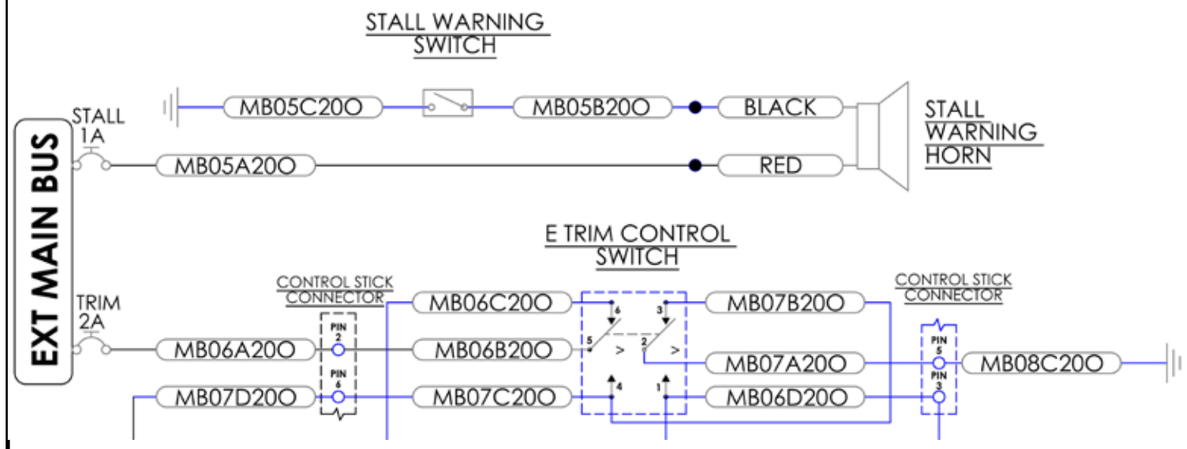
IGNITION BACKUP BATTERY SYSTEM



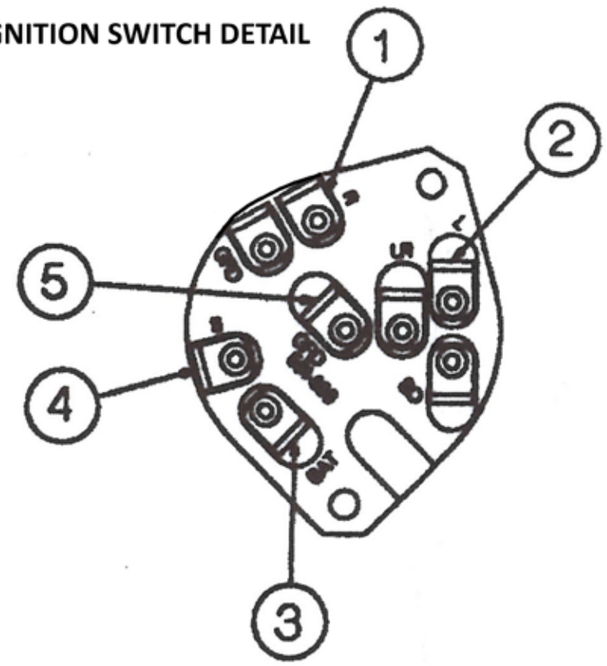
IGNITION SYSTEM



ALTERNATOR SYSTEM

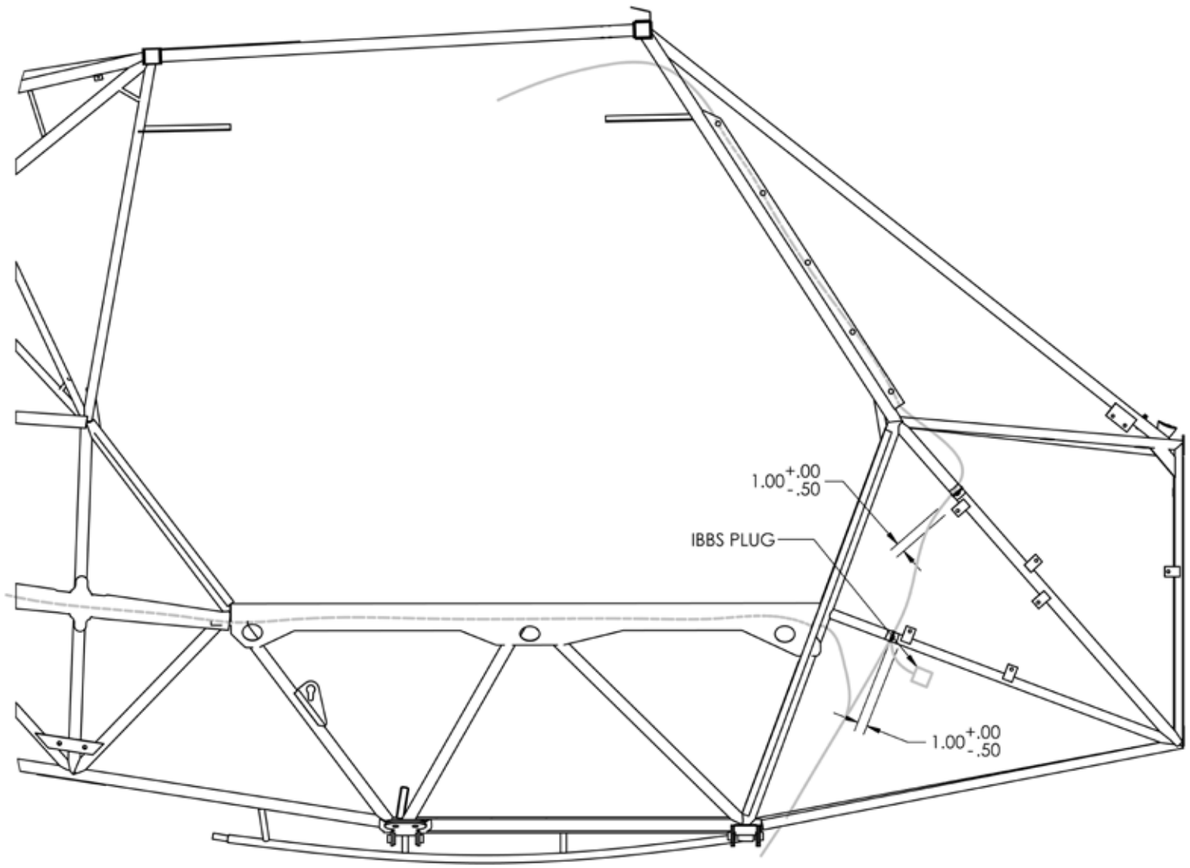


IGNITION SWITCH DETAIL

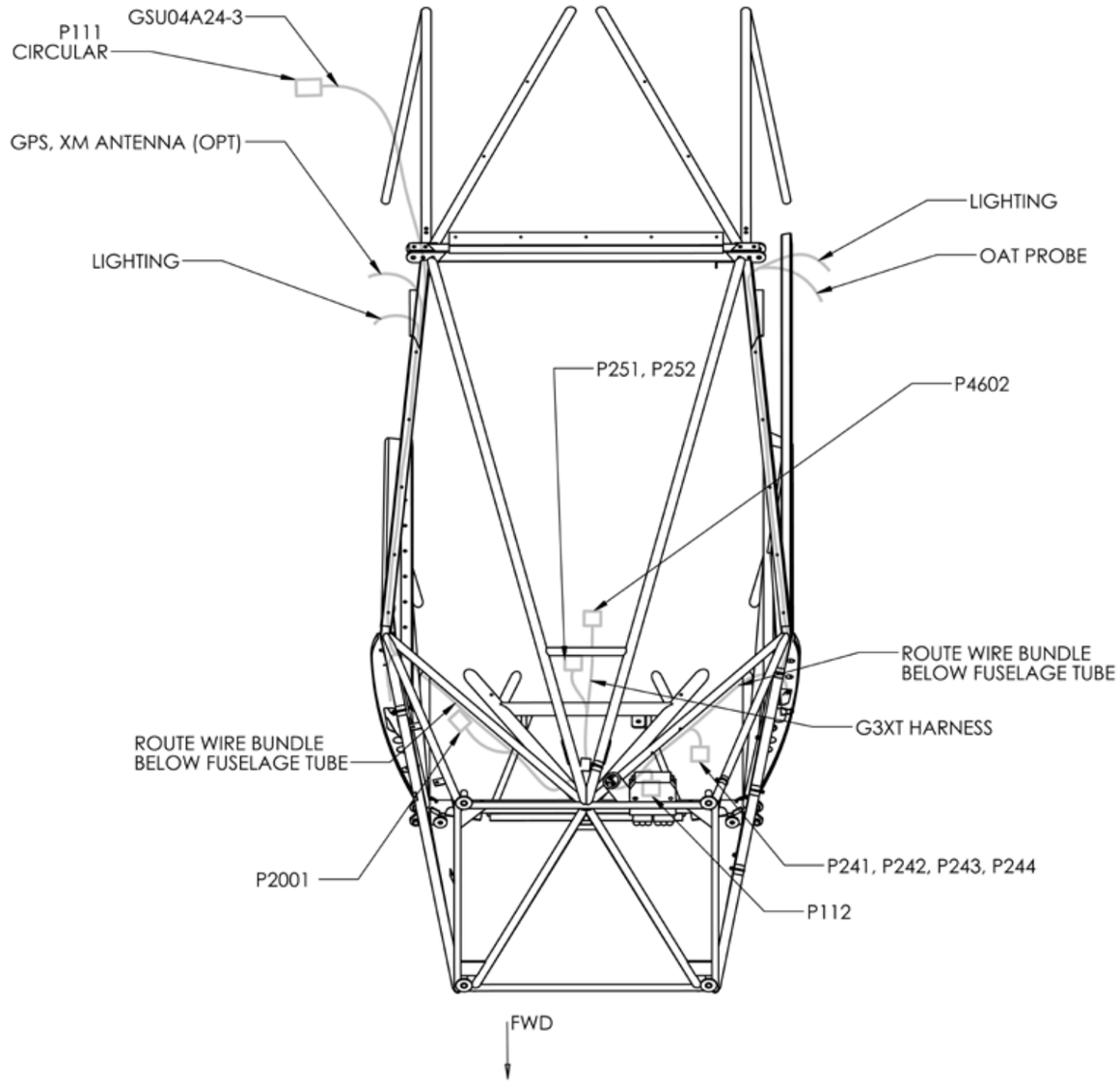


1. Connect wire from Left Ignition box to terminal 1.
2. Connect wire from Right Ignition box to terminal 2.
3. Connect Positive wire to terminal 3.
4. Connect Starter wire to terminal 4.
5. Connect terminal 5 (ground) as shown in ignition system diagram.

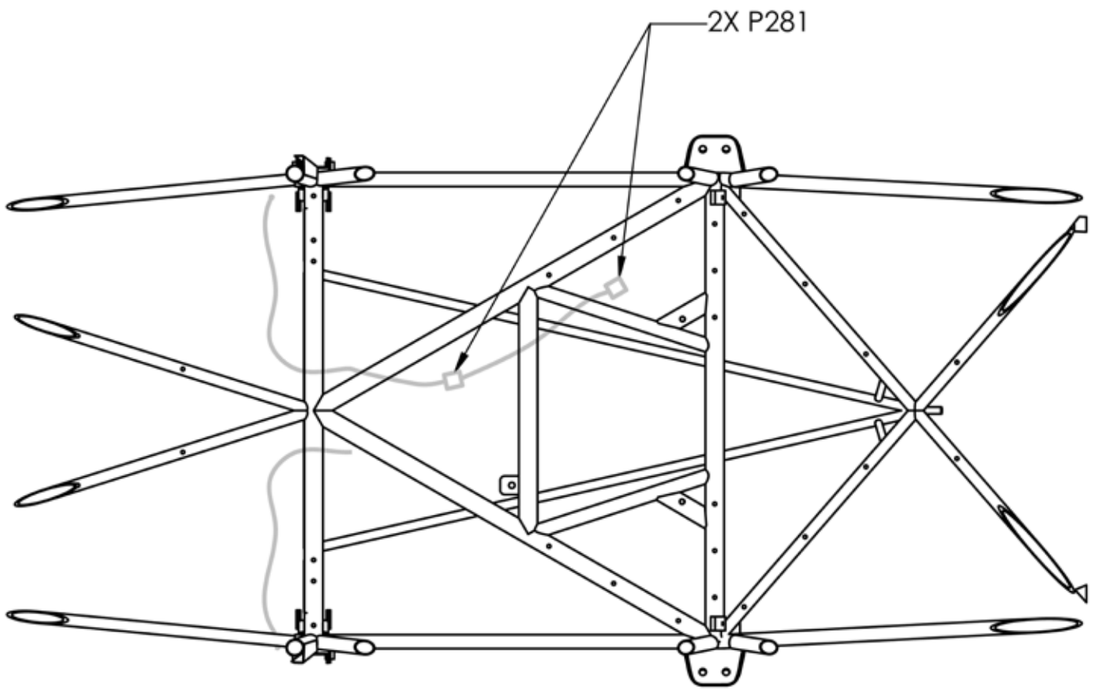
CONTROL STICK



GARMIN BACKUP BATTERY DETAIL

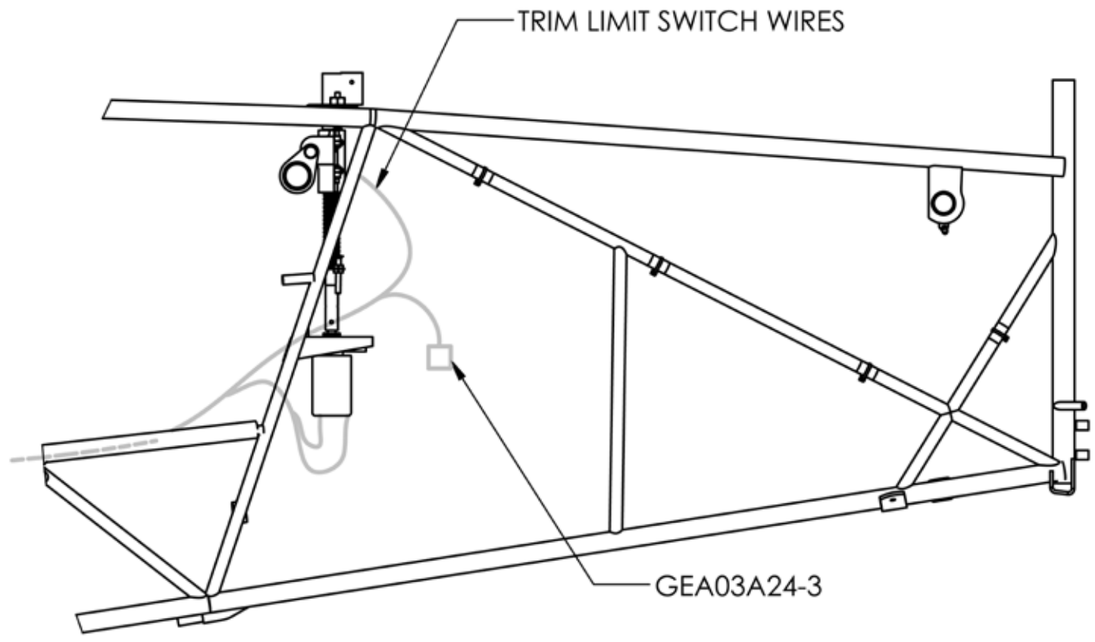


EXECUTIVE GLASS TOUCH PANEL
WIRE HARNESS INSTALLATION



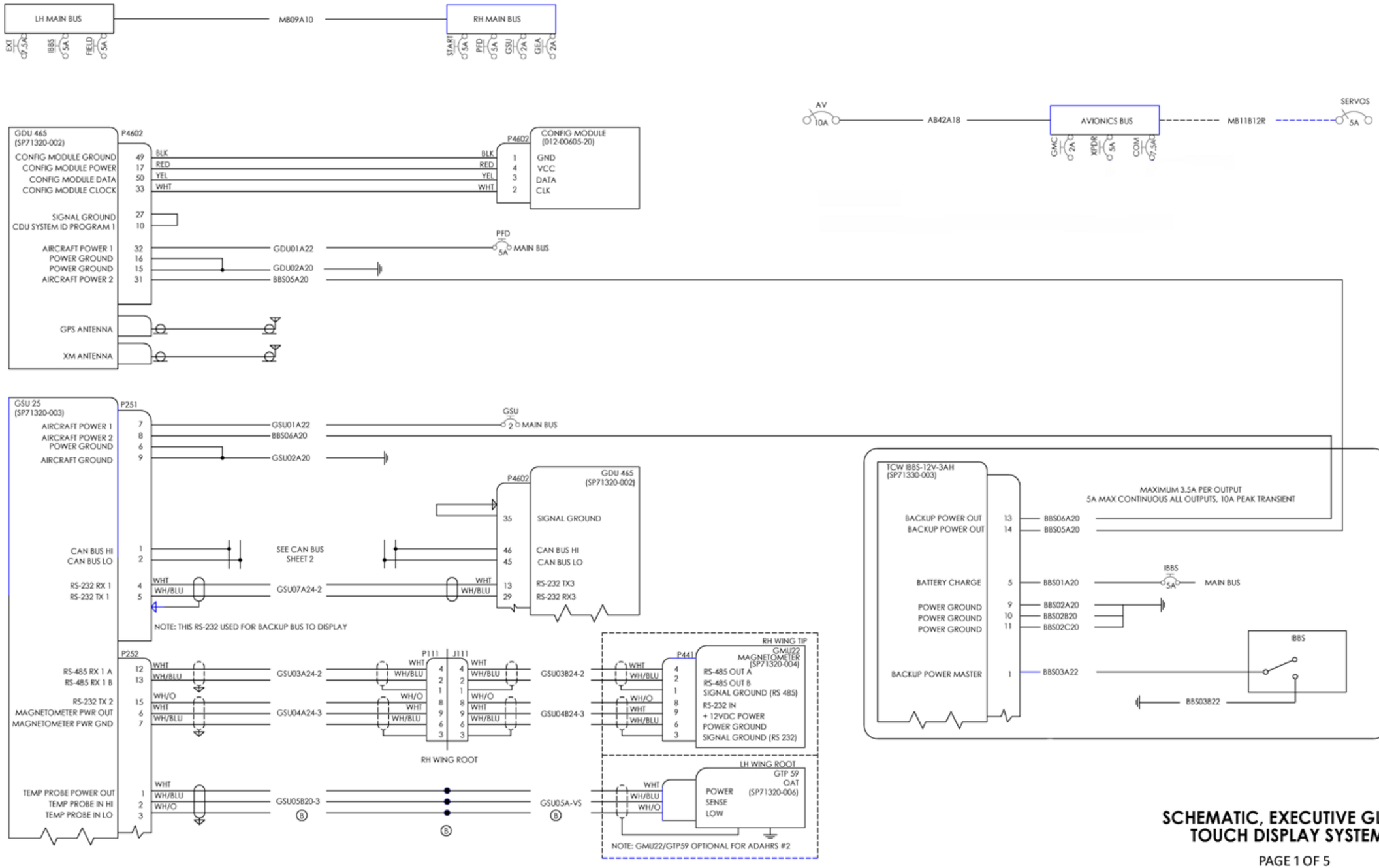
BELLY HARNESS WIRING DETAIL
VIEW LOOKING DOWN

FWD



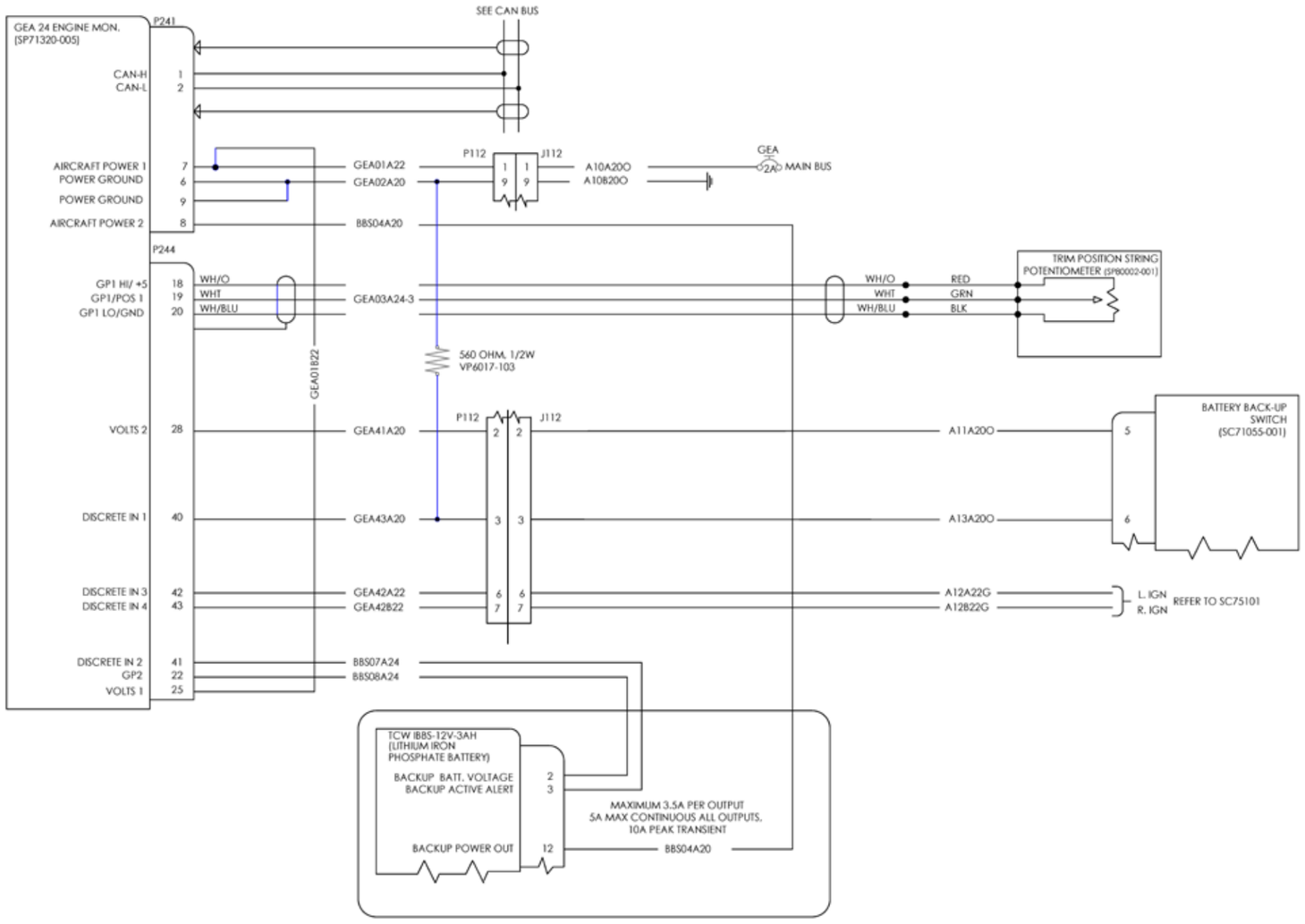
TRIM SENSOR WIRE HARNESS DETAIL

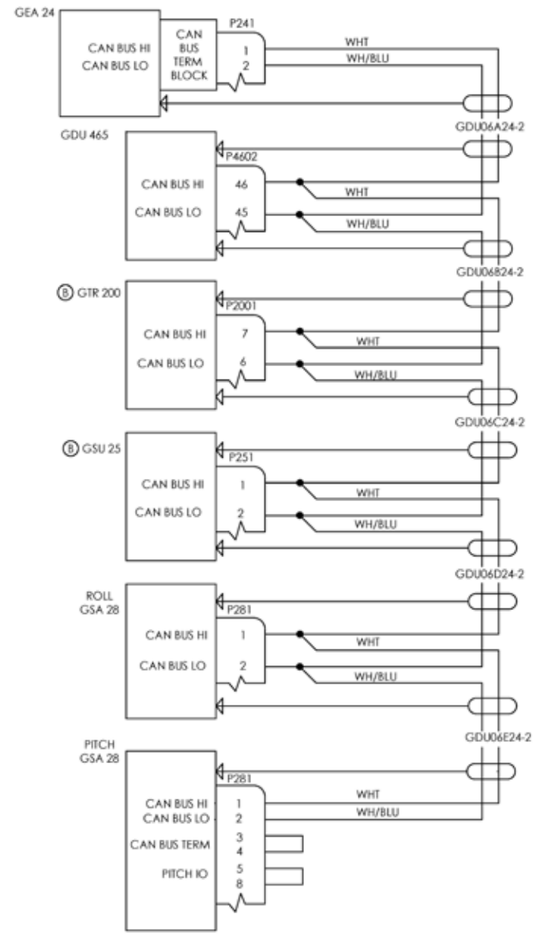
FWD



SCHEMATIC, EXECUTIVE GLASS TOUCH DISPLAY SYSTEM

PAGE 1 OF 5

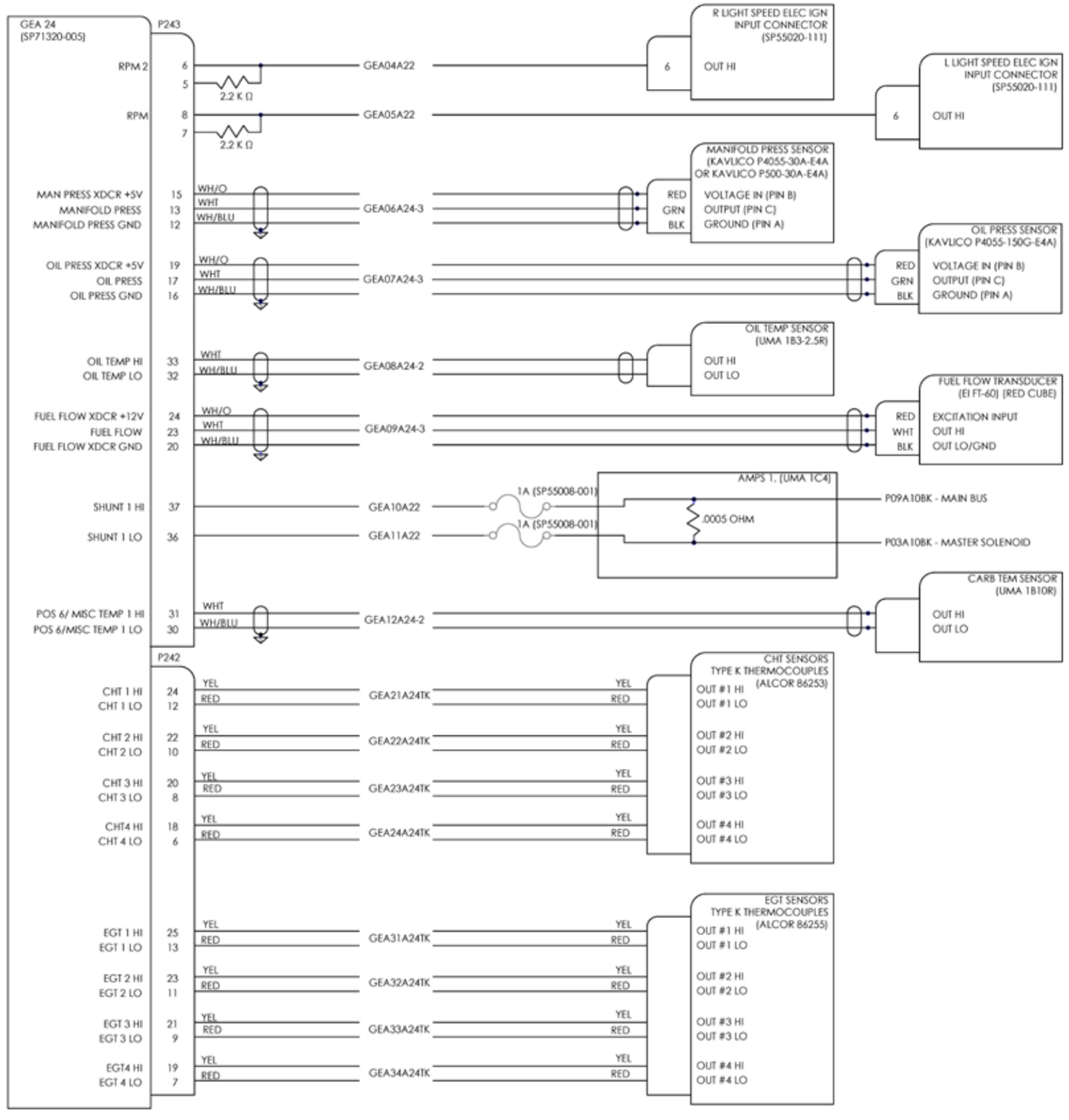




CAN BUS NETWORK

SCHEMATIC, EXECUTIVE GLASS TOUCH DISPLAY SYSTEM

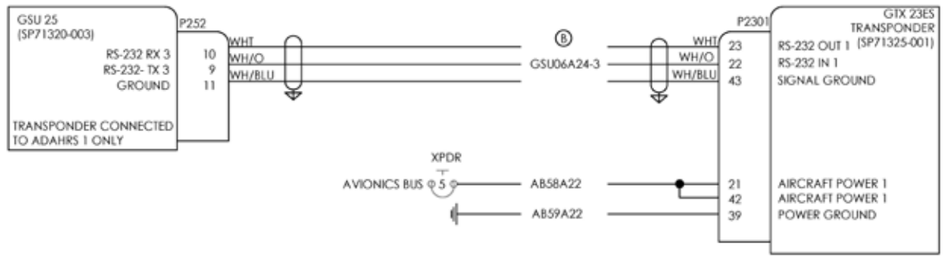
PAGE 2 OF 5



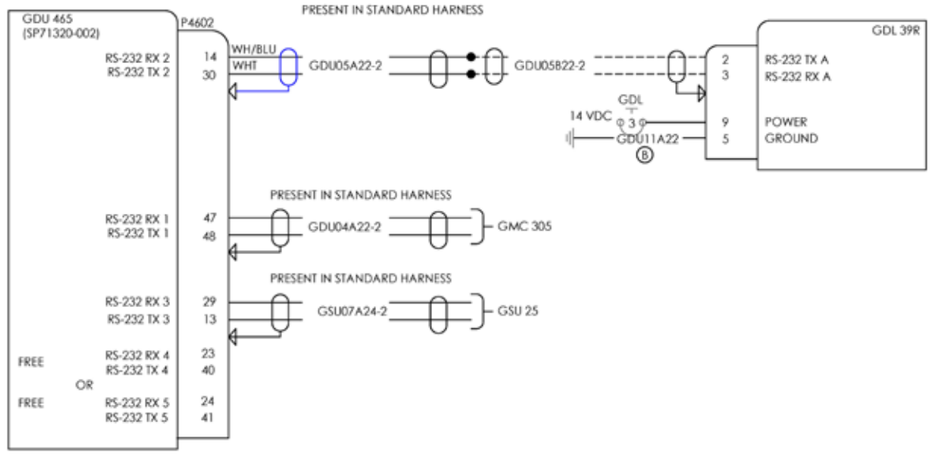
GEA 24-4/6 CYLINDER LYCOMING/CONTINENTAL SENSOR WIRING

SCHEMATIC, EXECUTIVE GLASS TOUCH DISPLAY SYSTEM

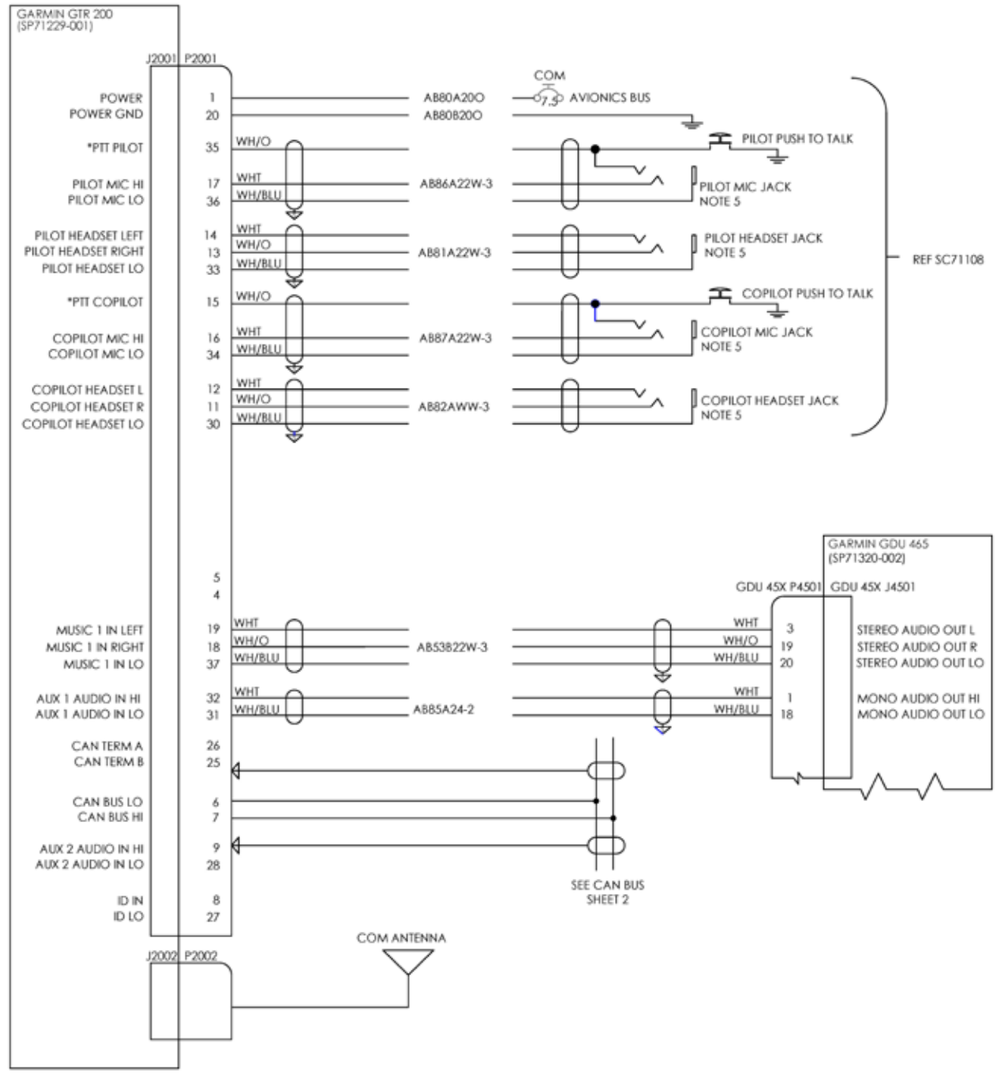
PAGE 3 OF 5



GSU 25 - GTX 23ES TRANSPONDER INTERCONNECT/CONFIGURATION



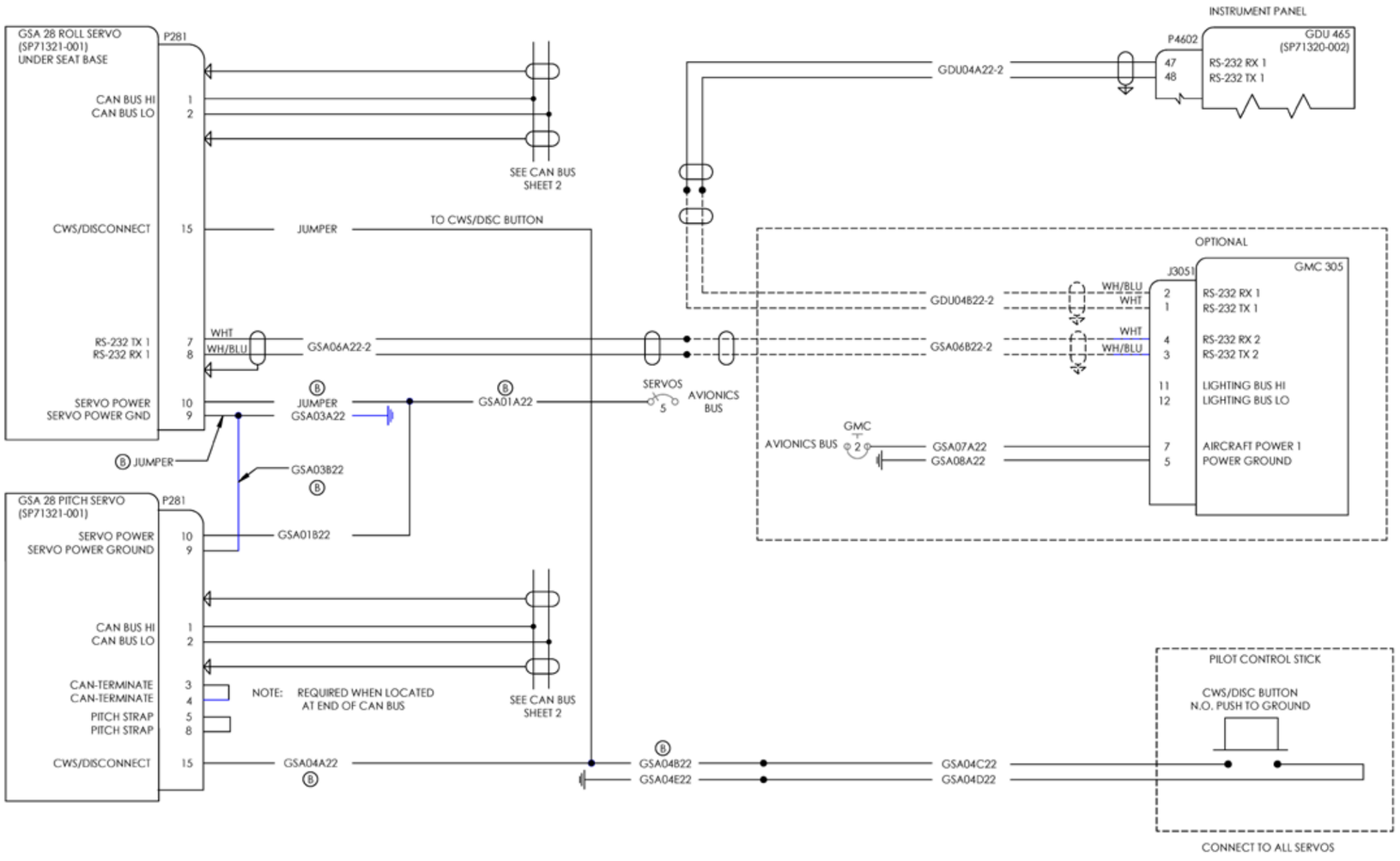
GDL 39R INTERCONNECT/CONFIGURATION
(OPTIONAL)



GTR 200/GDU 465 INTERCONNECT DRAWING

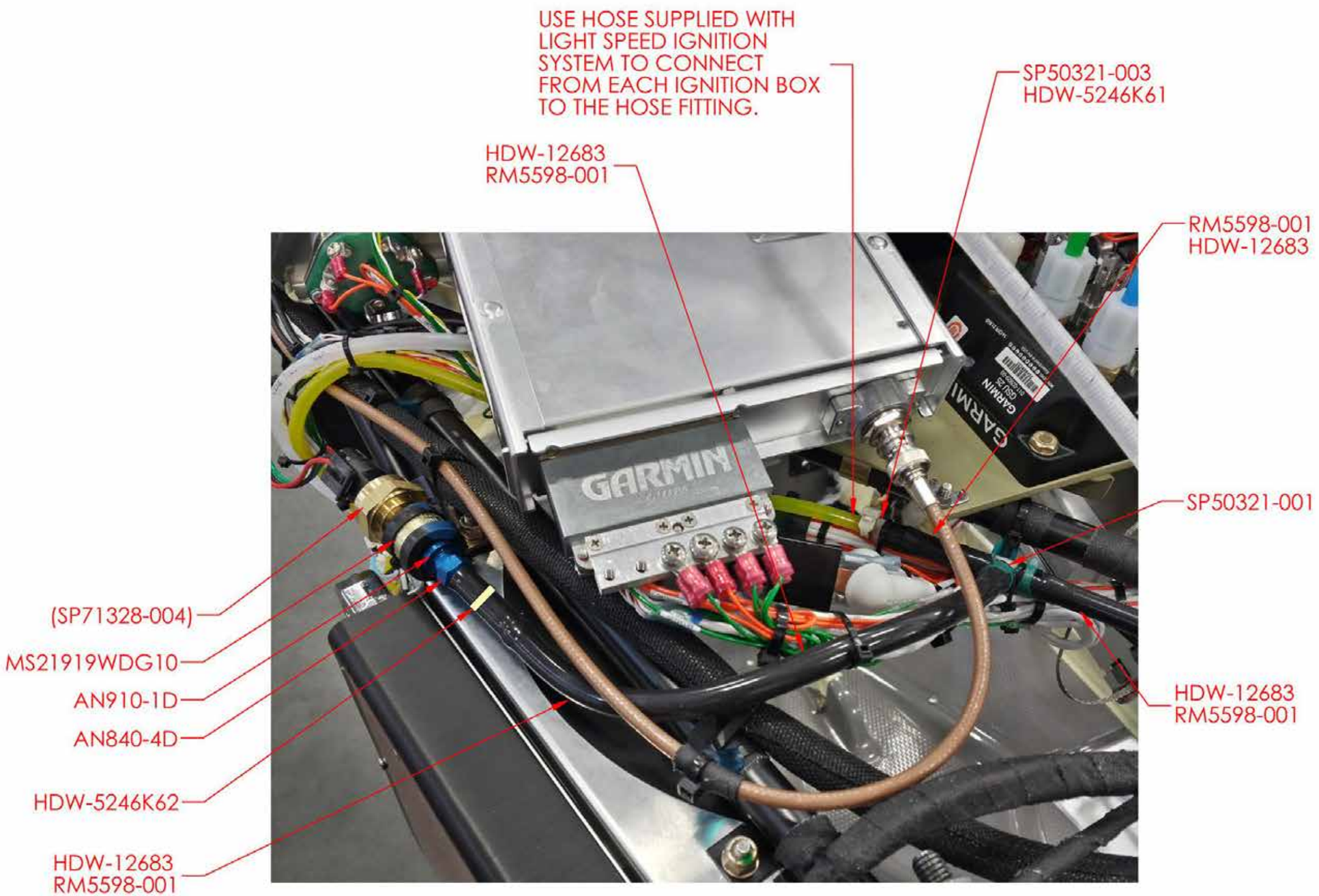
SCHEMATIC, EXECUTIVE GLASS TOUCH DISPLAY SYSTEM

PAGE 4 OF 5

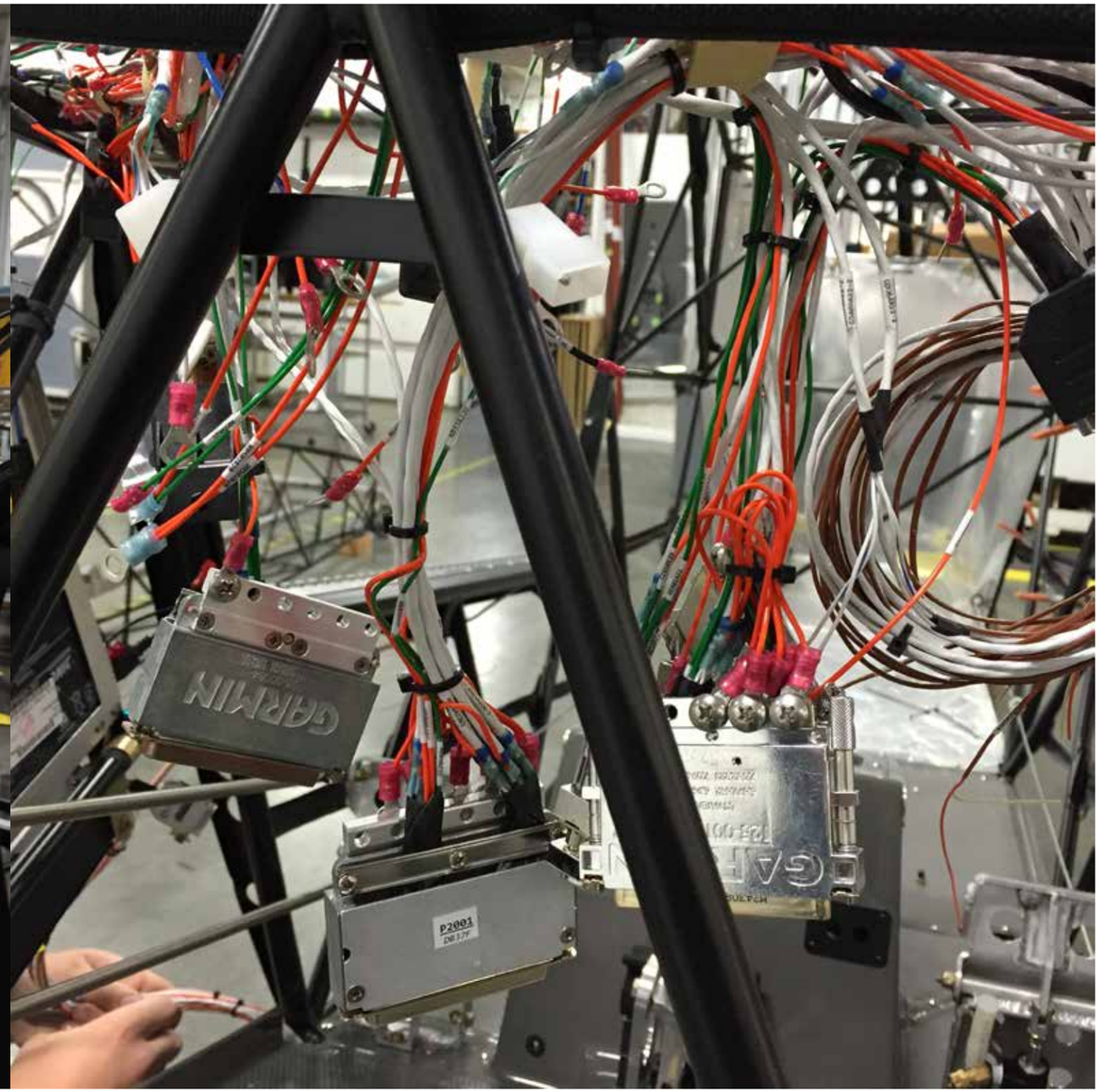


SCHEMATIC, EXECUTIVE GLASS TOUCH DISPLAY SYSTEM

PAGE 5 OF 5



MANIFOLD PRESSURE SENSOR



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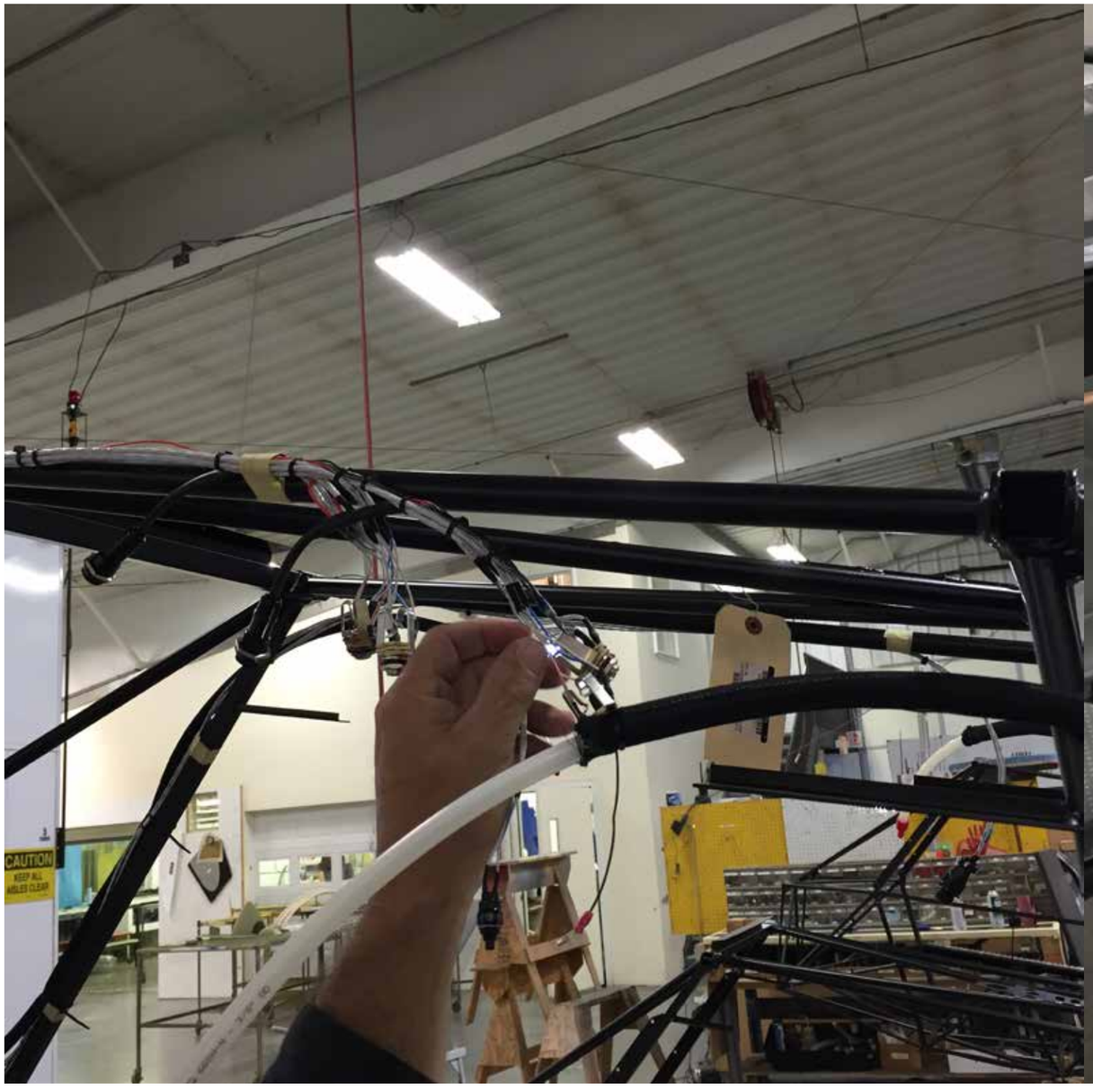
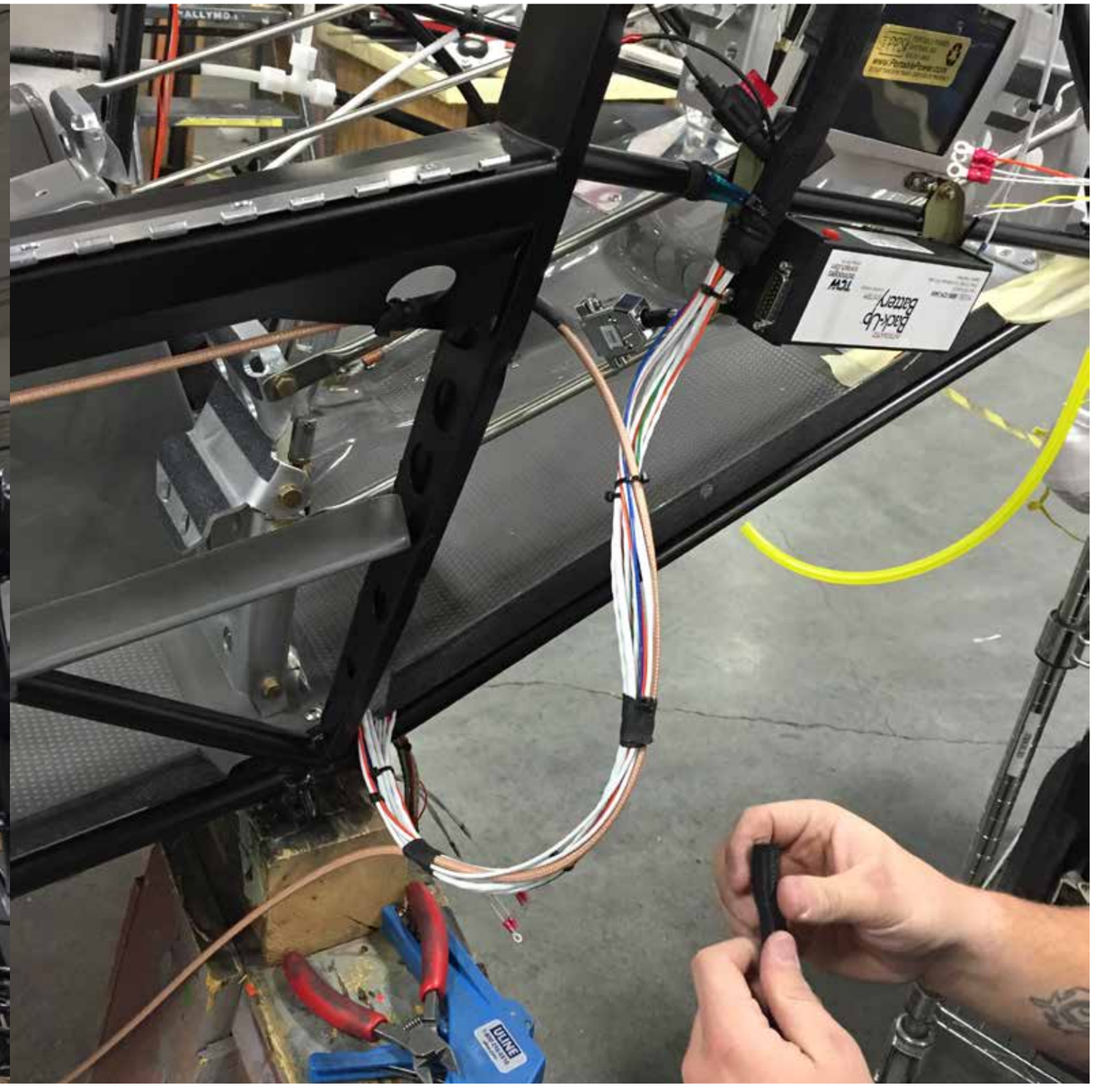
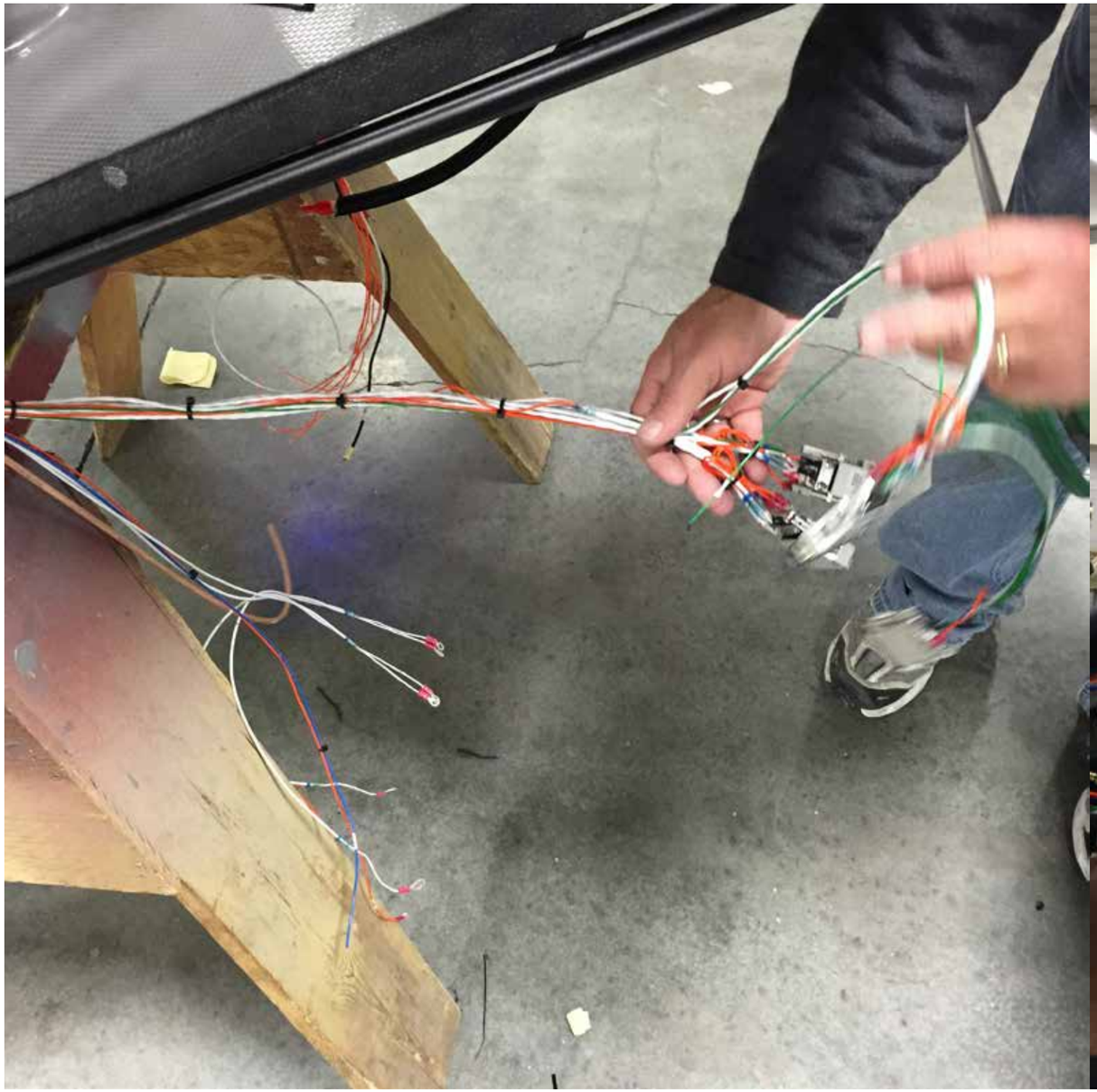
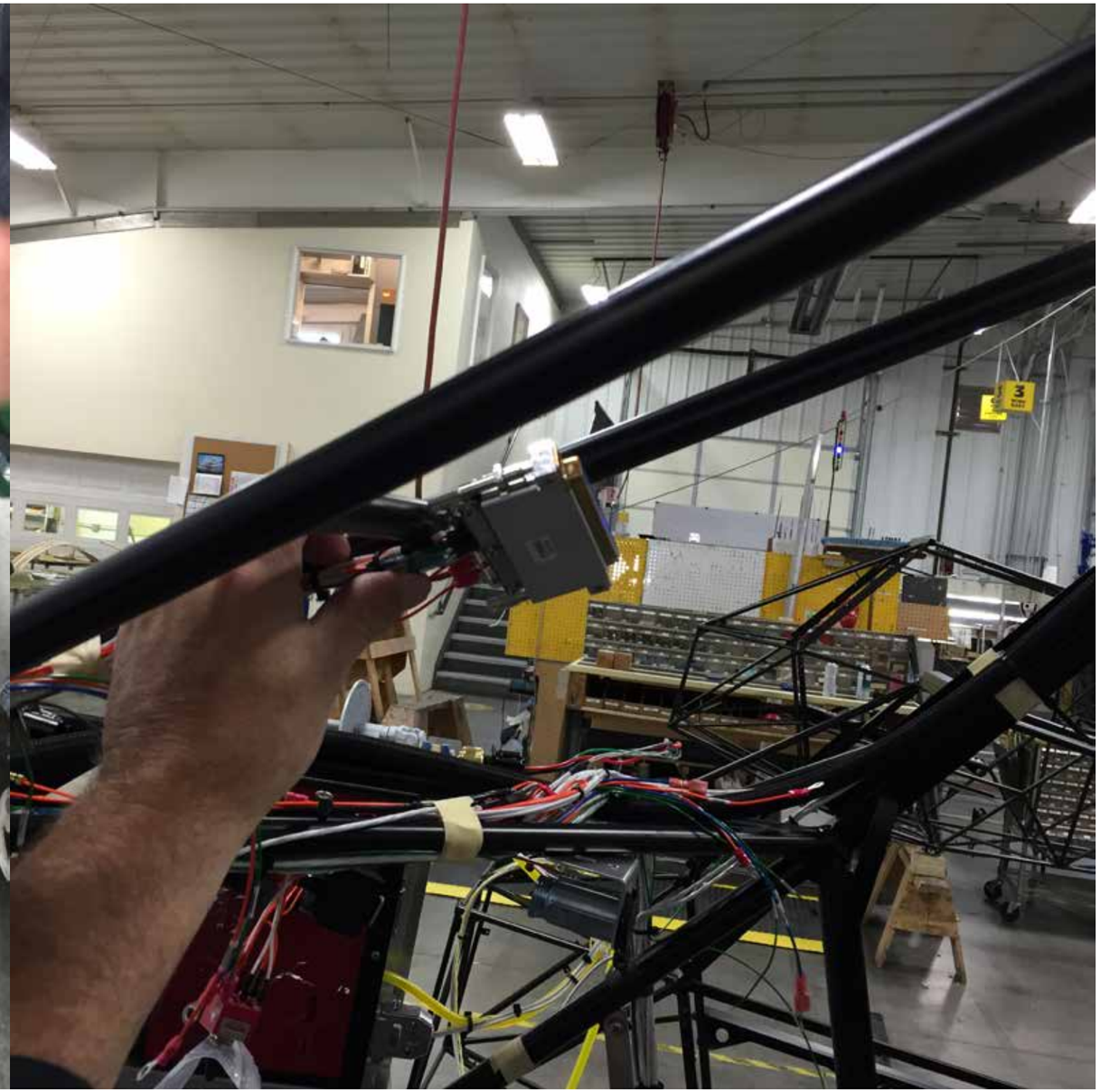
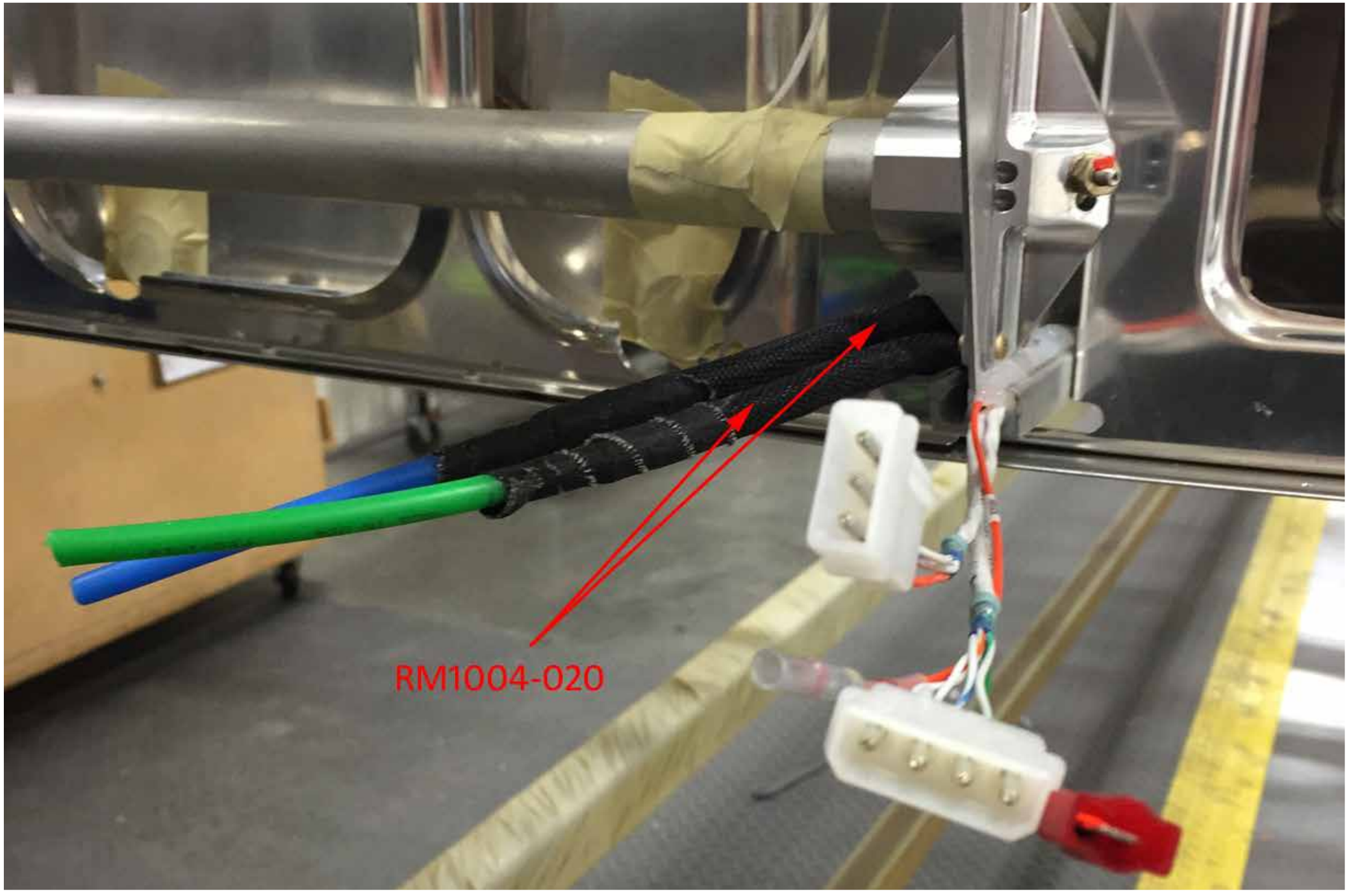
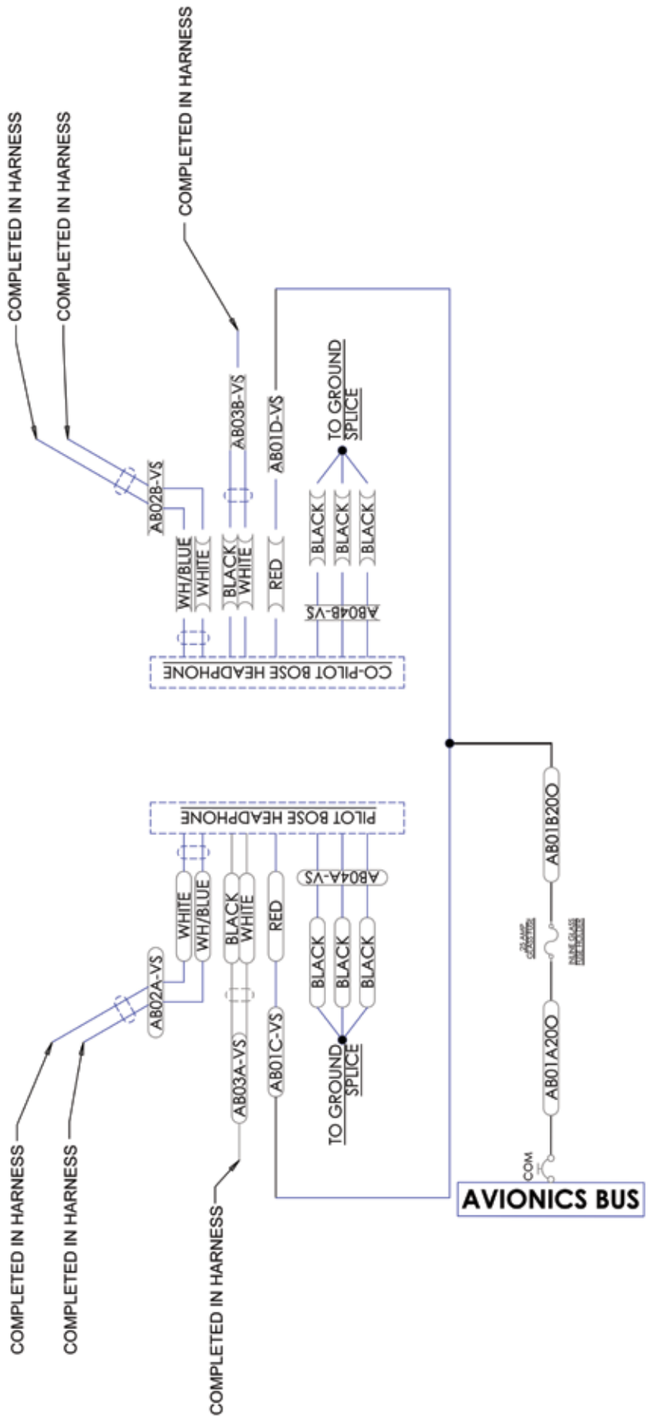












BOSE HEADPHONE CONNECTION SCHEMATIC

WIRE CODE LOGIC EXAMPLE

WIRE CODE: **A01A18TR-3**

A	01	A	18	TR	-3
SYSTEM DESCRIPTION CODE	CIRCUIT NUMBER	CIRCUIT SEGMENT	WIRE SIZE (AWG)	COLOR CODE	SEE NOTES

5

NOTES:

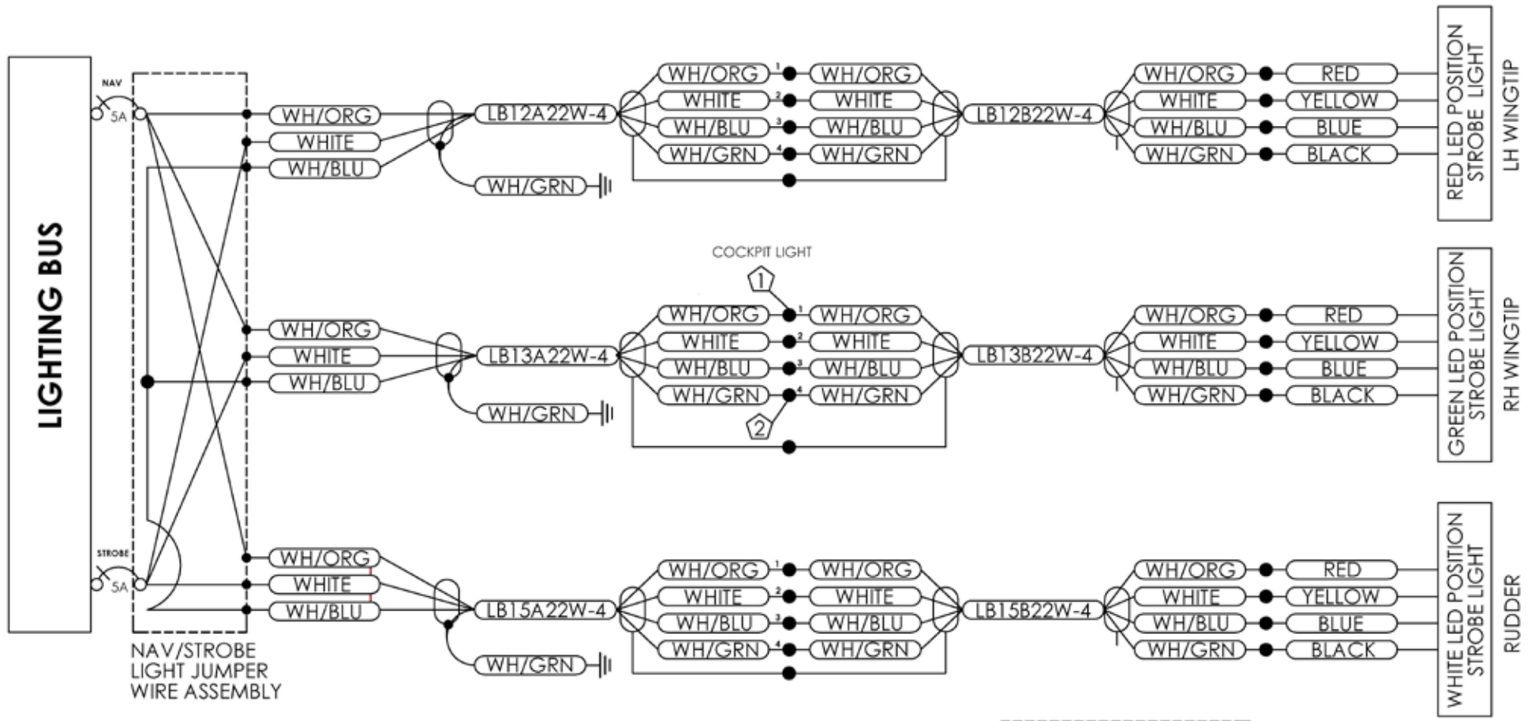
1. 'VS' AFTER THE DASH DENOTES VENDOR SUPPLIED WIRING
2. VENDOR SUPPLIED WIRING DOES NOT ADHERE TO COLOR/AWG/MATERIAL RELATIONSHIP
3. IF A NUMBER EXISTS AFTER THE DASH THEN THE WIRE IS SHIELDED.
4. ALL AWG WIRES ARE COPPER
5. SEE AWG-COLOR-MATERIAL RELATIONSHIP

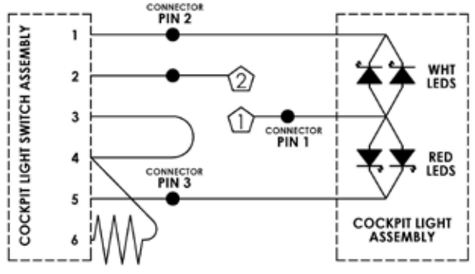
SYSTEM DESCRIPTION

CODE	DESCRIPTION
A	ALTERNATOR
AB	AVIONICS BUS
IG	IGNITION
LB	LIGHTING BUS
MB	MAIN BUS
P	POWER

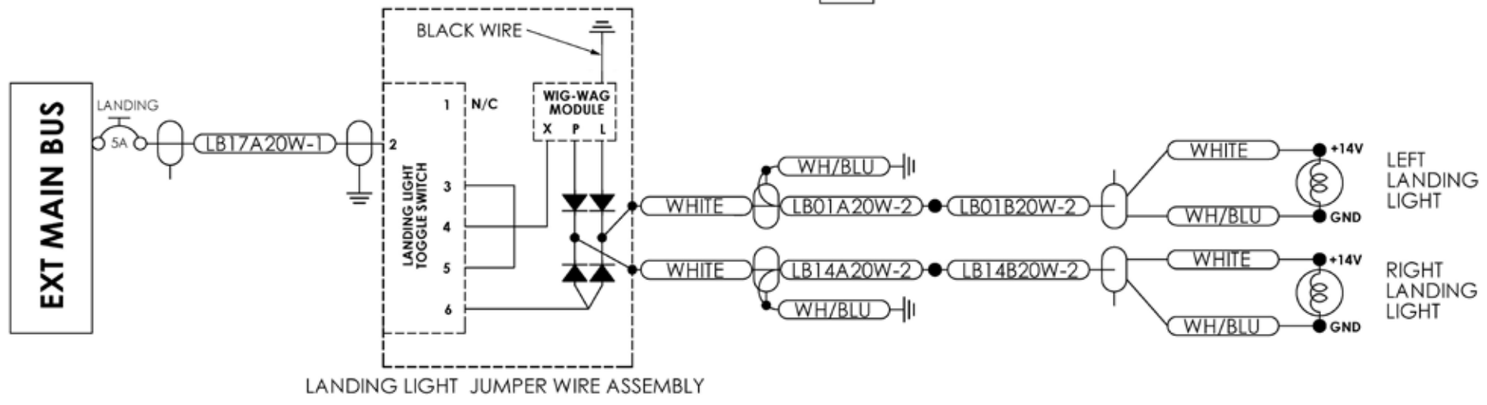
AWG/COLOR/MATERIAL RELATIONSHIP

CODE	AWG-COLOR-CONDUCTOR	MATERIAL PART NUMBER
04W	04 AWG - WHITE	HDW-22759-16-4
08W	08 AWG - WHITE	HDW-22759-16-8
10BK	10 AWG - BLACK	HDW-22759-16-10
12R	12 AWG - RED	HDW-22759-16-12
14W	14 AWG - WHITE	HDW-22759-16-14-9
16BN	16 AWG - BROWN	HDW-22759-16-16
18BL	18 AWG - BLUE	HDW-22759-16-18
18W-1	18 AWG SHIELDED - 1 CONDUCTOR - WHITE	HDW-27500/18-18-1
20W-1	20 AWG SHIELDED - 1 CONDUCTOR - WHITE	HDW-27500/20-20-1
20O	20 AWG - ORANGE	HDW-22759-16-20
22G	22 AWG - GREEN	HDW-22759-16-22
22W-2	22 AWG SHIELDED - 2 CONDUCTOR - WHITE	HDW-27500/22-22-2
22W-3	22 AWG SHIELDED - 3 CONDUCTOR - WHITE	HDW-27500/22-22-3
18TR-3	18 AWG SHIELDED - 3 CONDUCTOR - TRANSLUCENT RED	88770
142B	COAX RG-142	RG-142





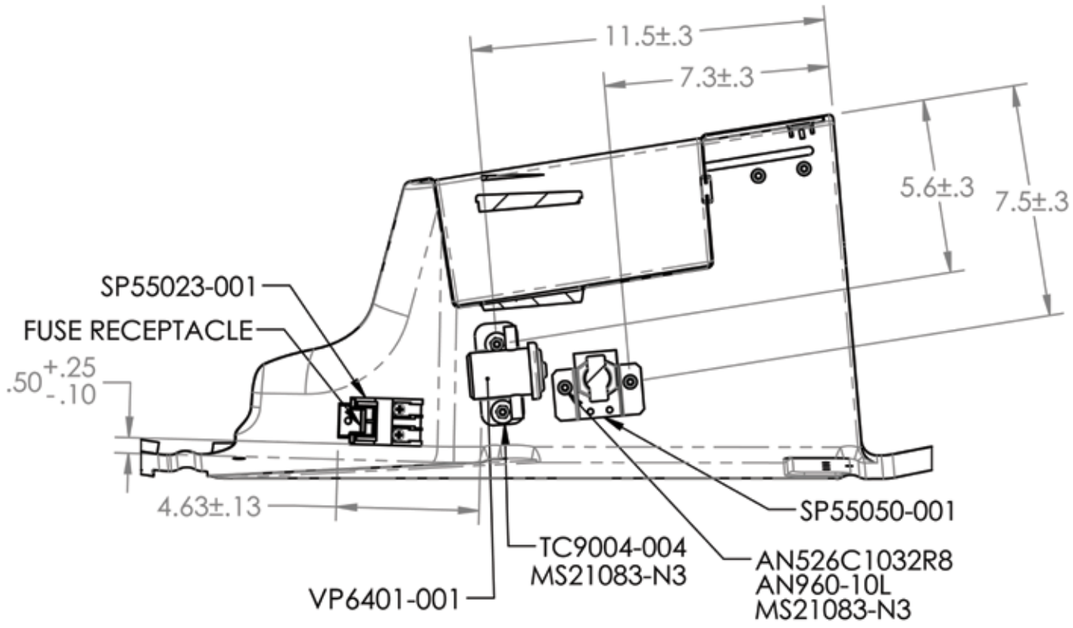
COCKPIT LIGHT WIRING DETAIL

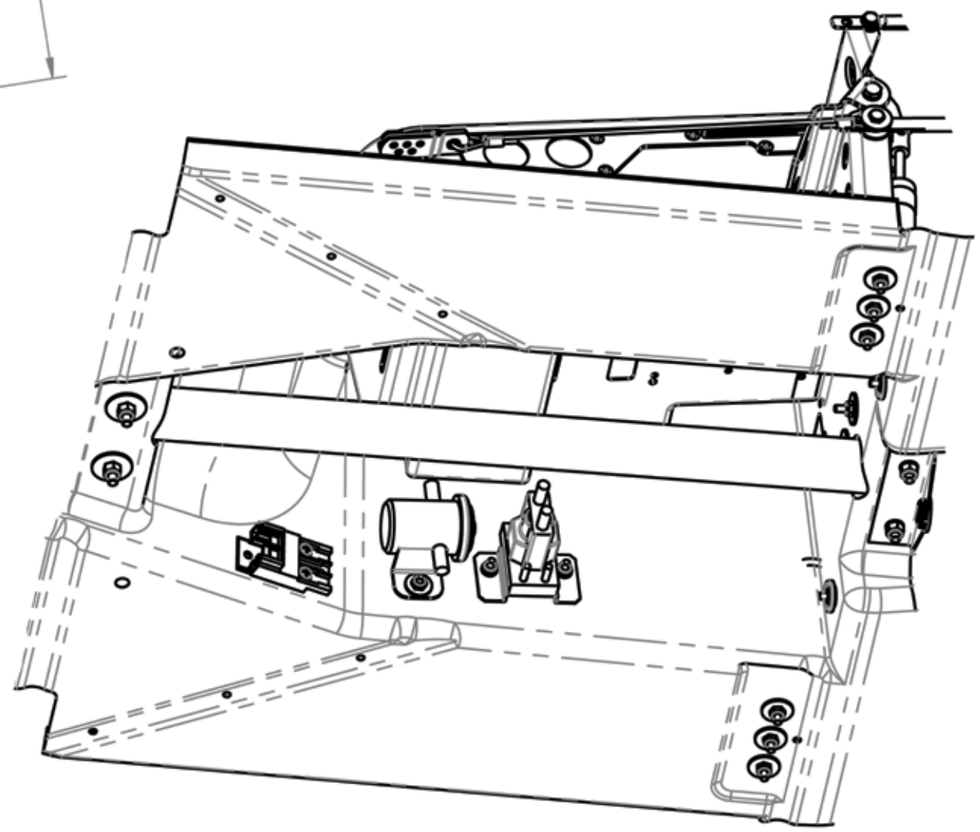


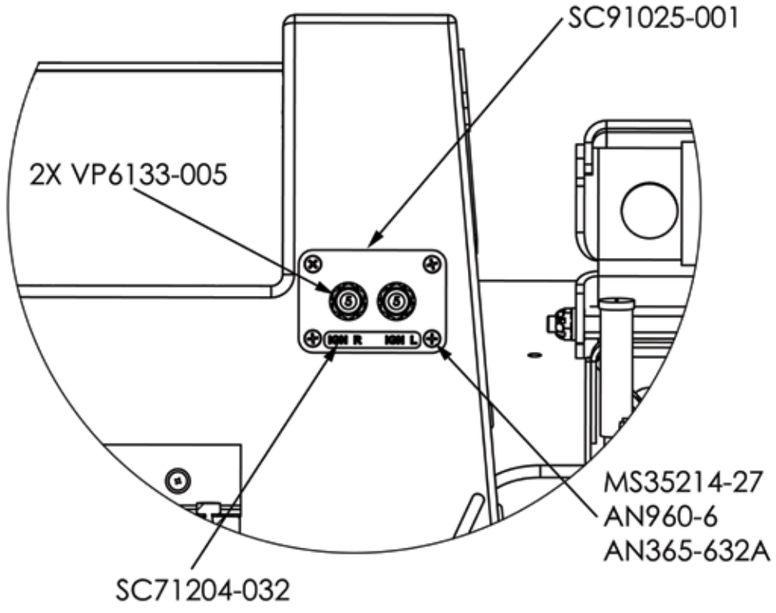
LANDING LIGHT CIRCUIT

NOTES:

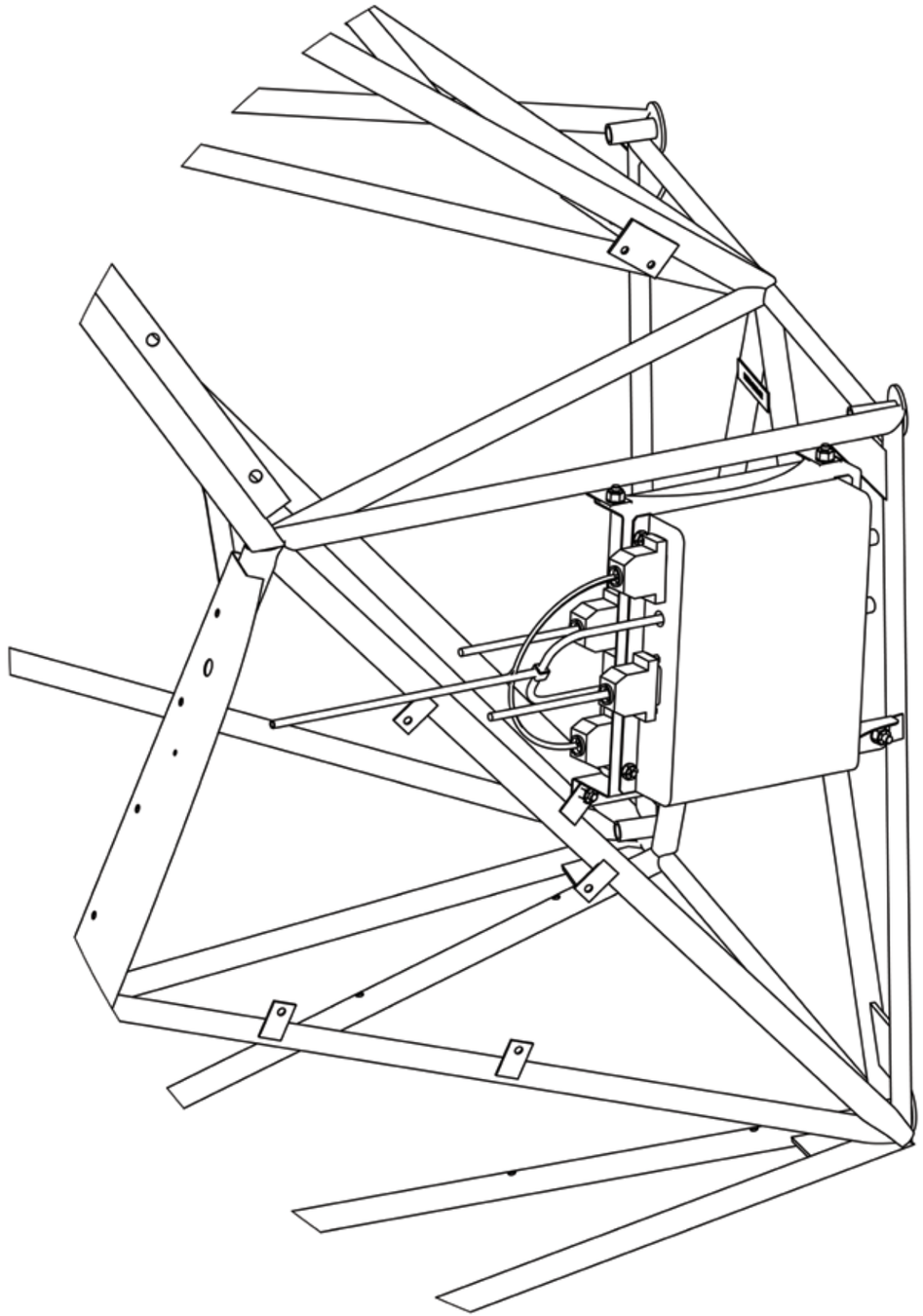
1. WIRES CROSSING WITHOUT DOT ARE NOT CONNECTED.
2. REFERENCE WIRE CODE LOGIC, SC76101-001.

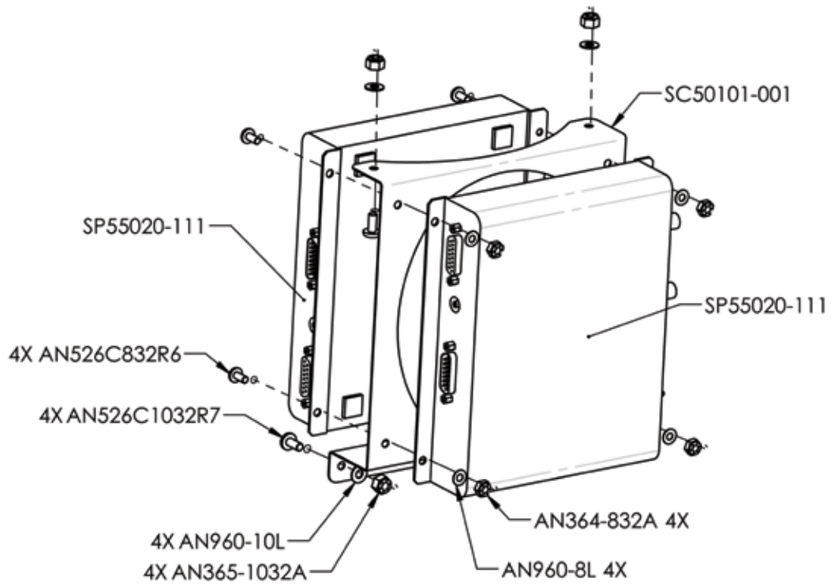




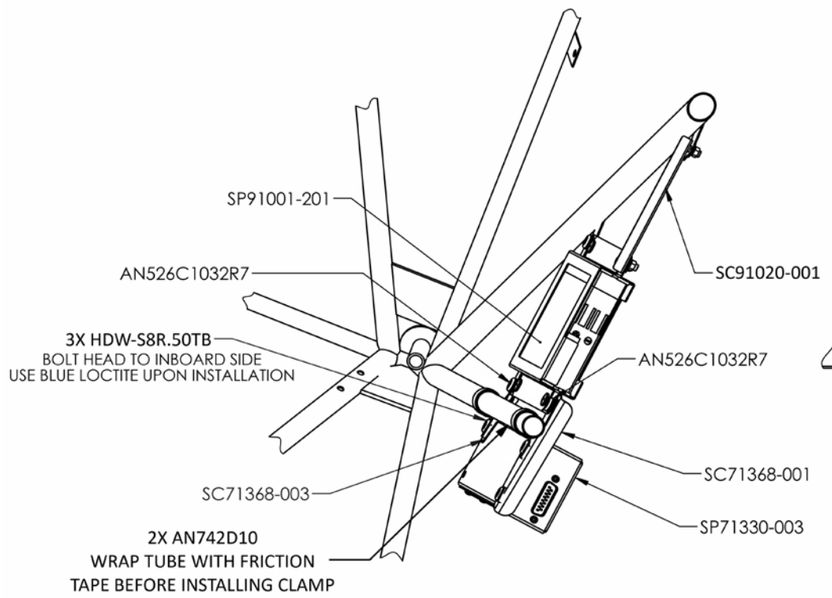


SOLENOID AND BREAKER PLATE INSTALLATION

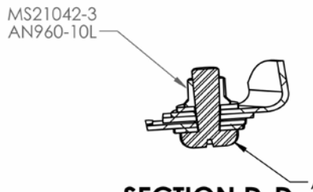




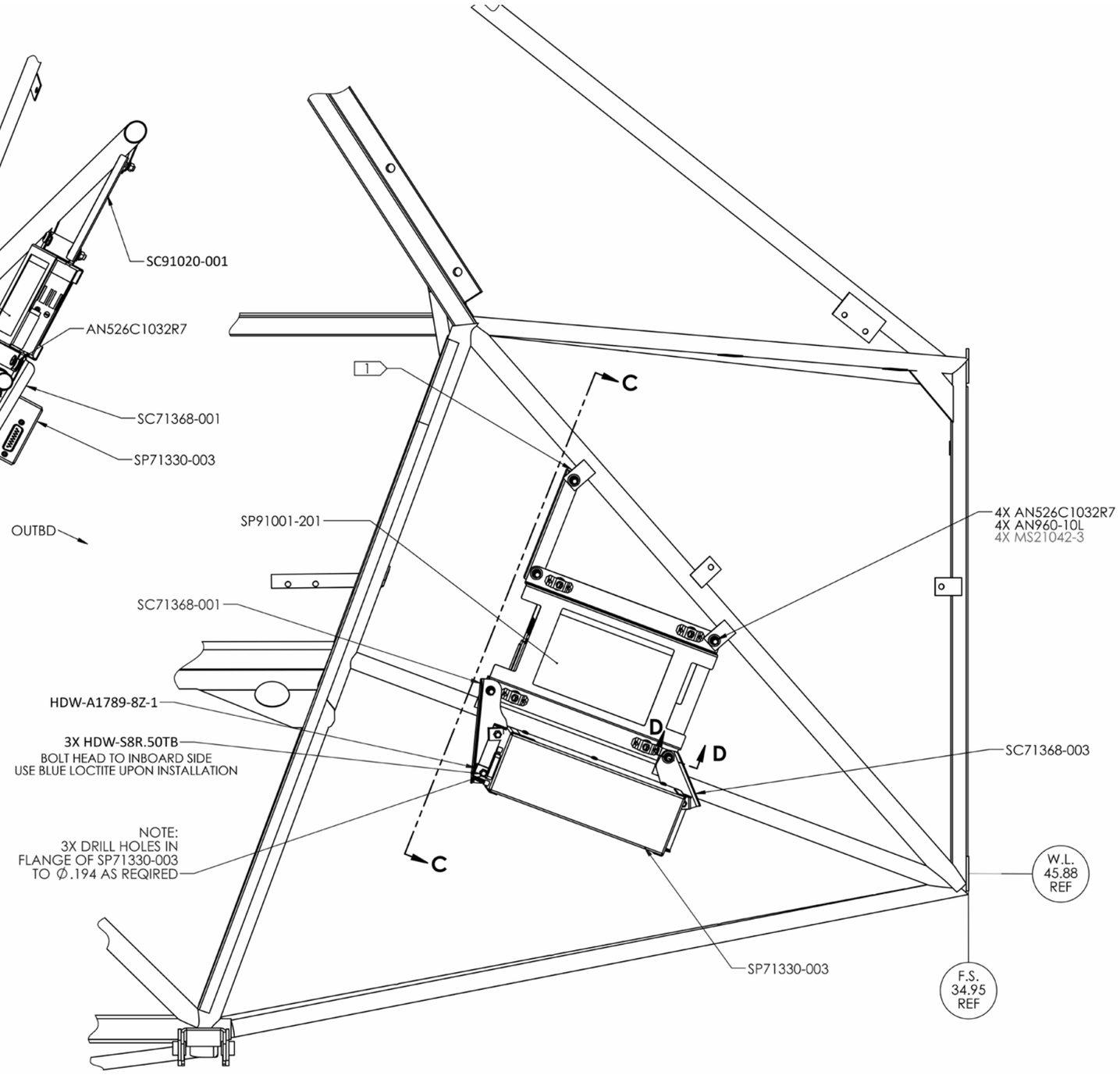
IGNITION MODULES HARDWARE DETAIL

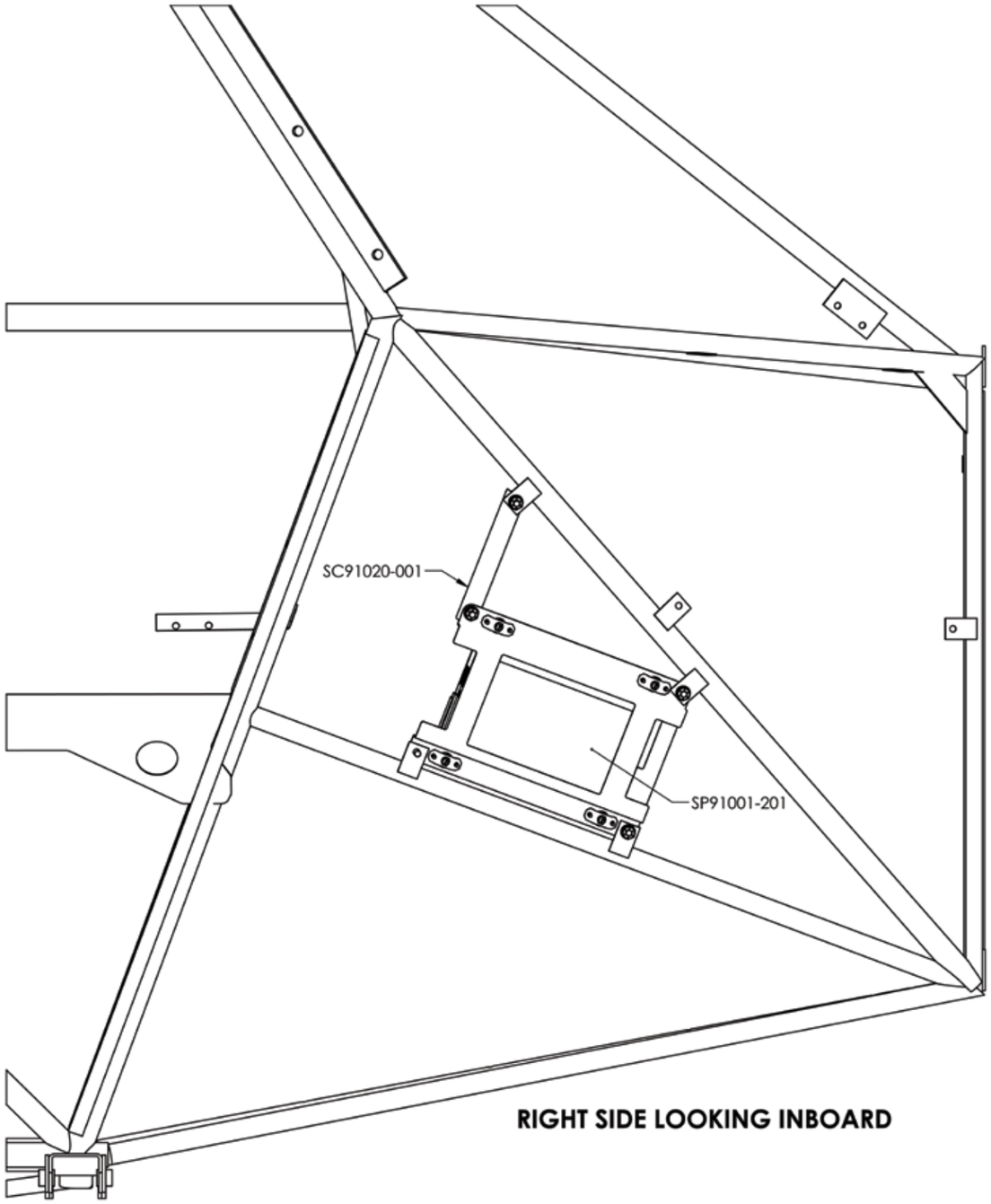


SECTION C-C
VIEW LOOKING FWD

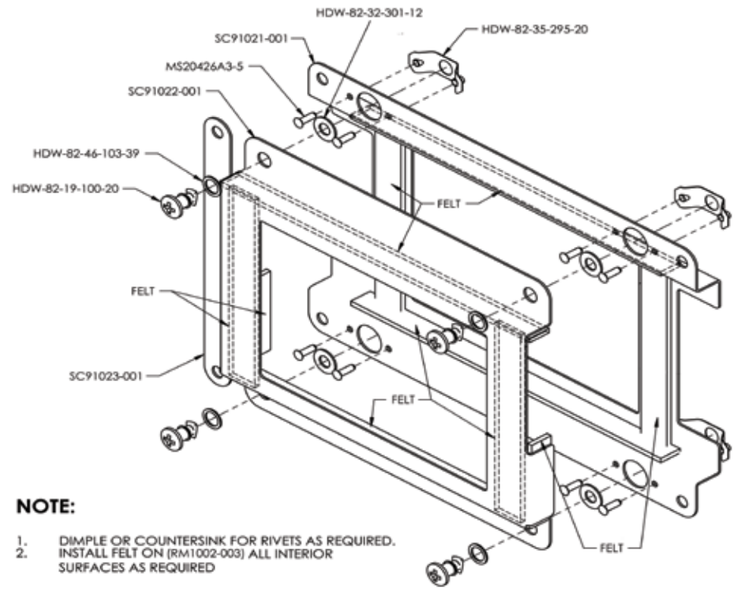


SECTION D-D
TYPICAL
(ROTATED 32.5° CCW)



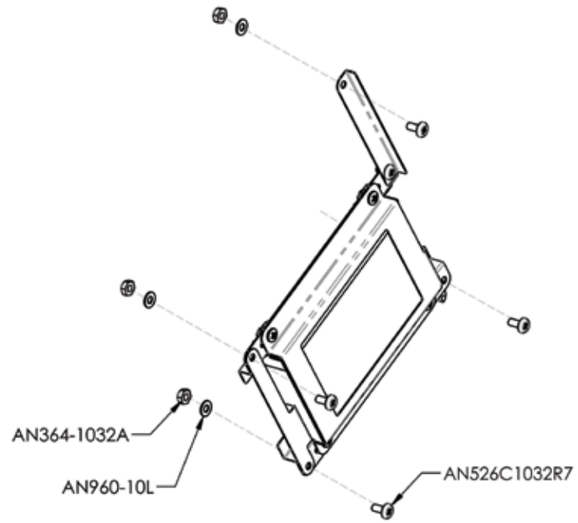


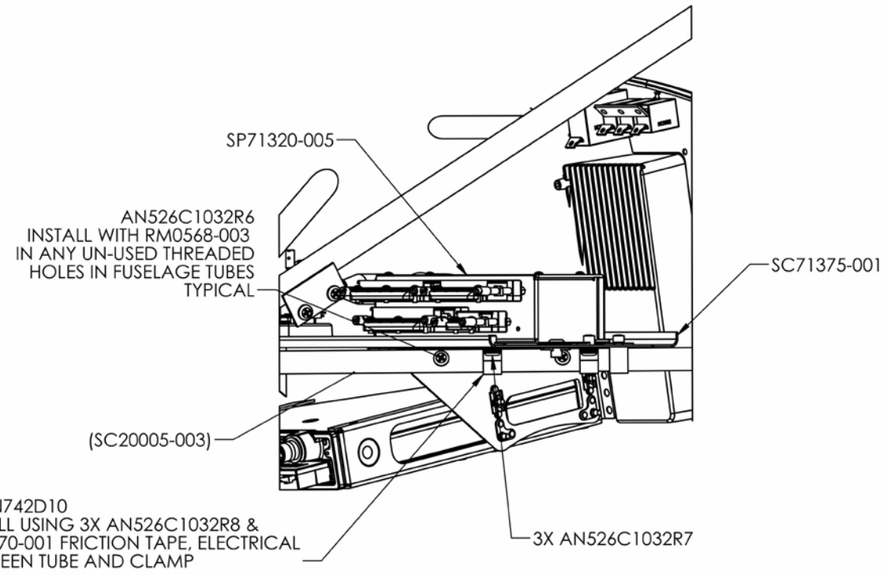
RIGHT SIDE LOOKING INBOARD

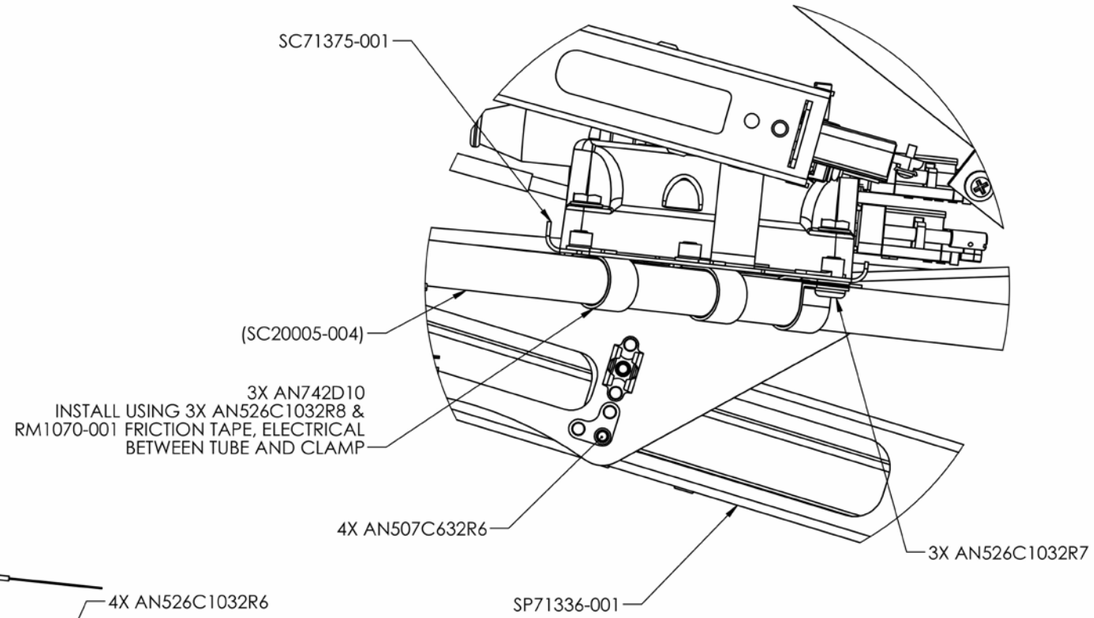


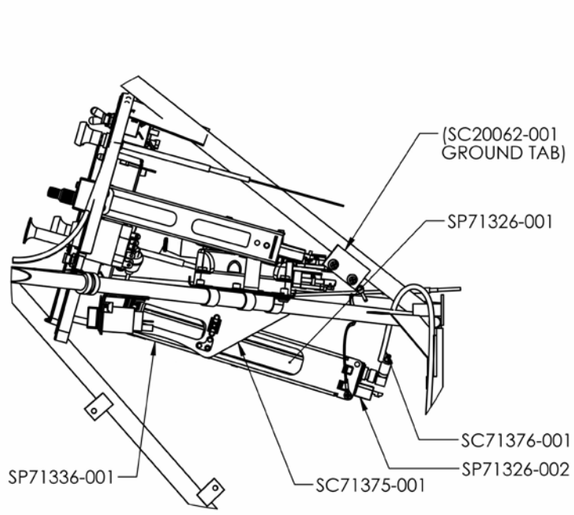
NOTE:

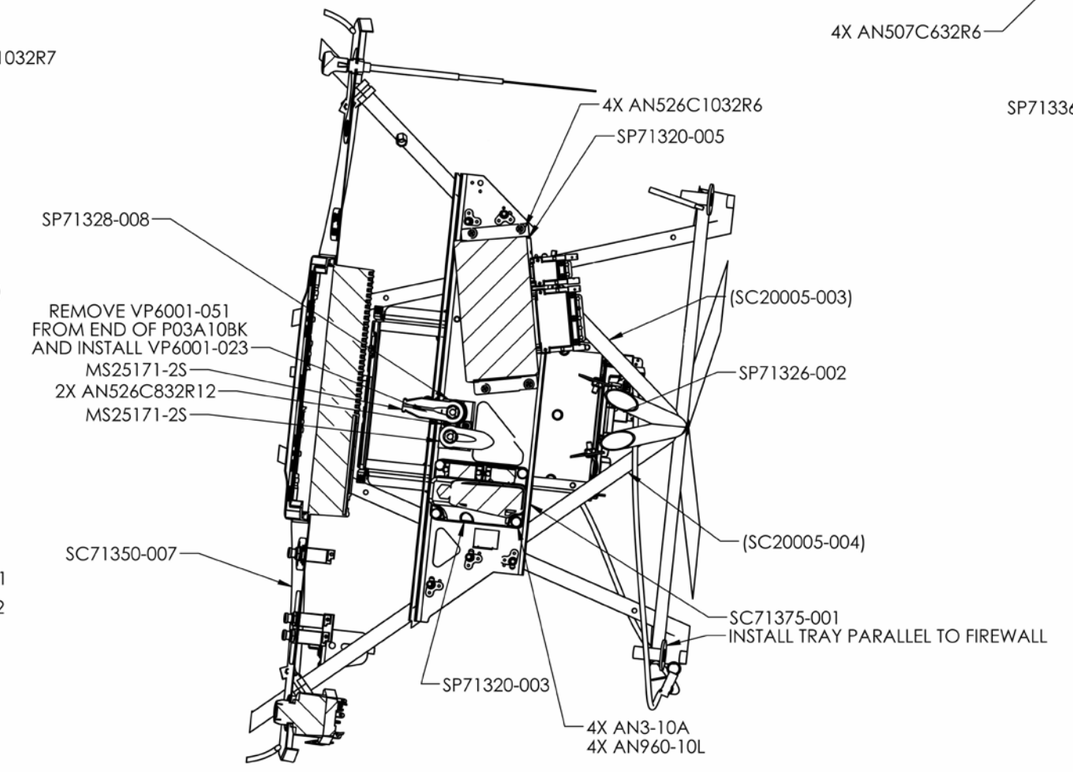
1. DIMPLE OR COUNTERSINK FOR RIVETS AS REQUIRED.
2. INSTALL FELT ON (RM1002-003) ALL INTERIOR SURFACES AS REQUIRED

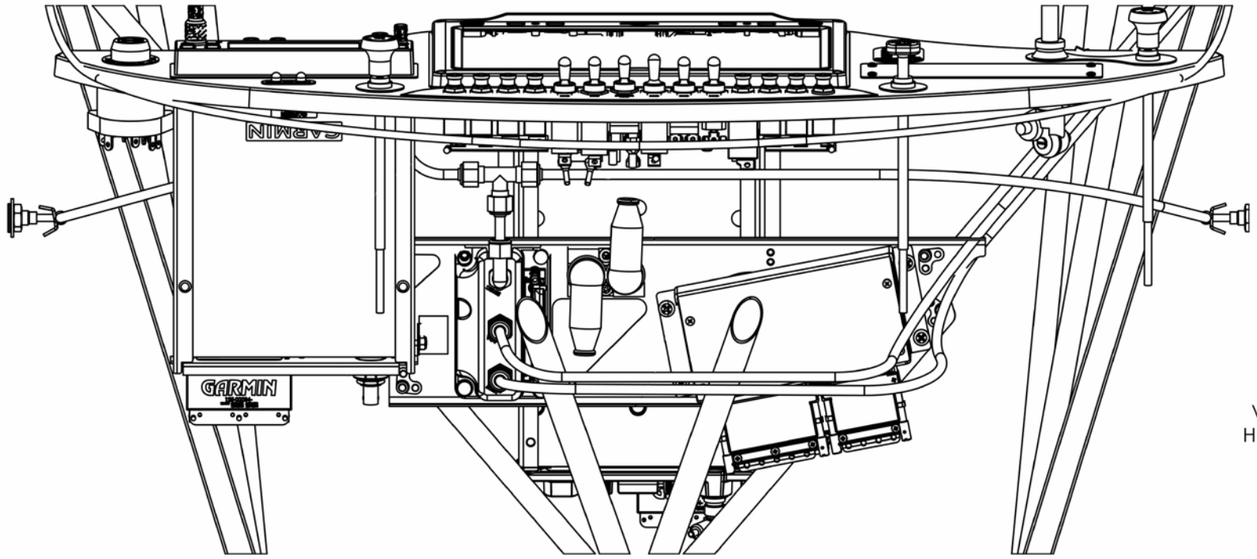


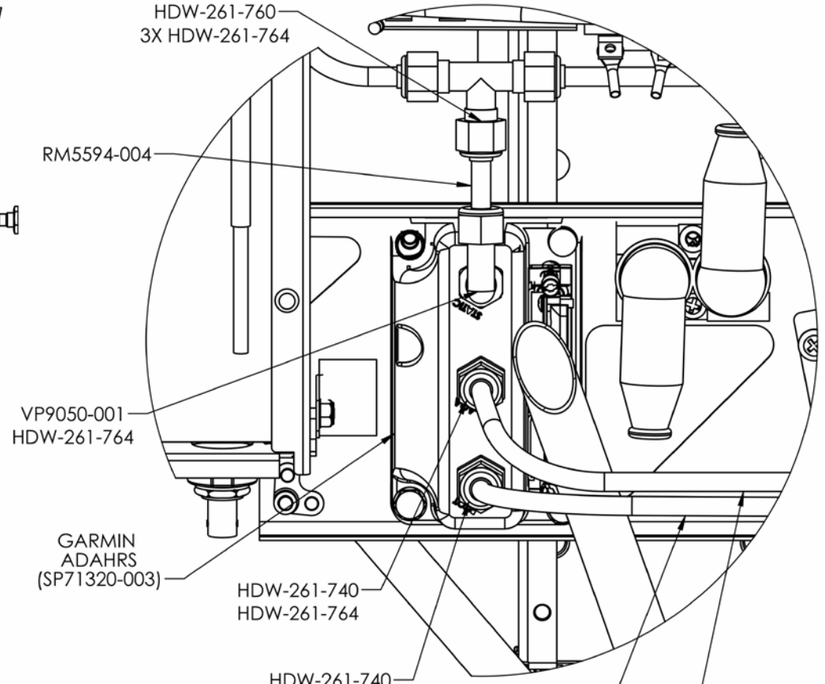


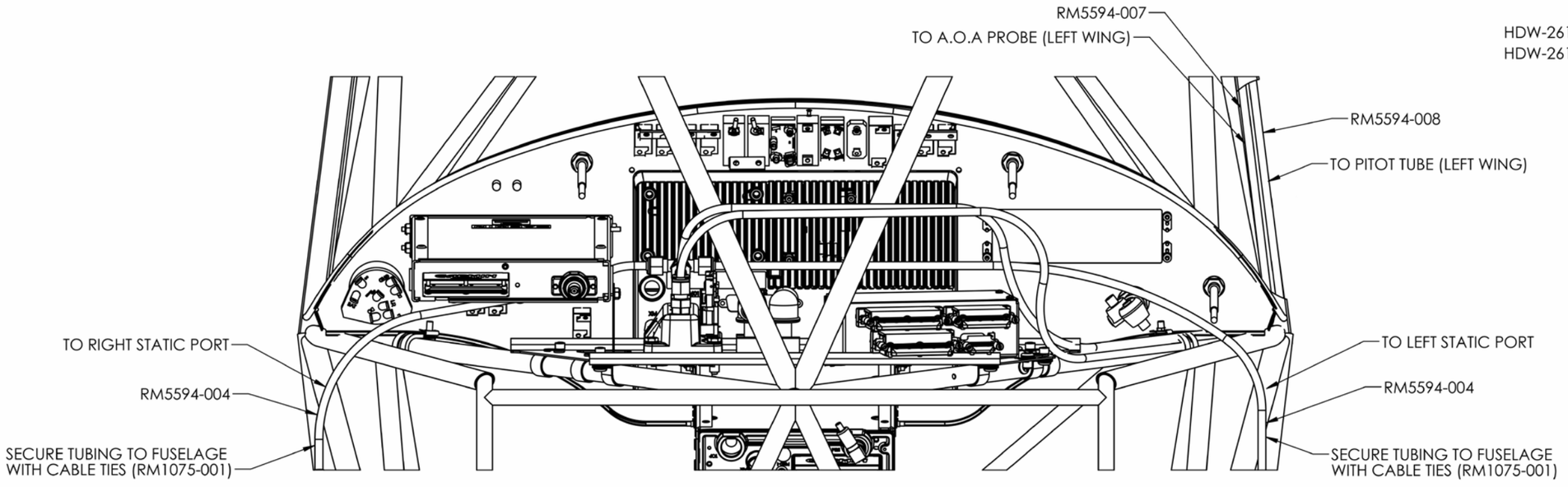


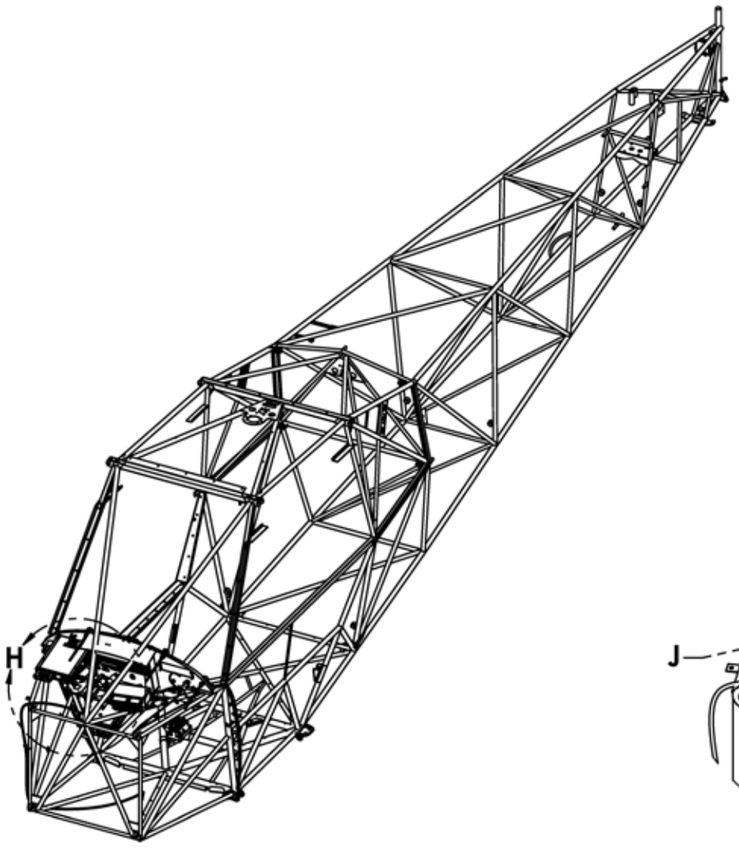


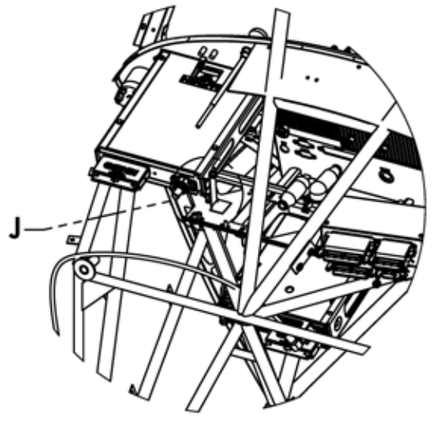




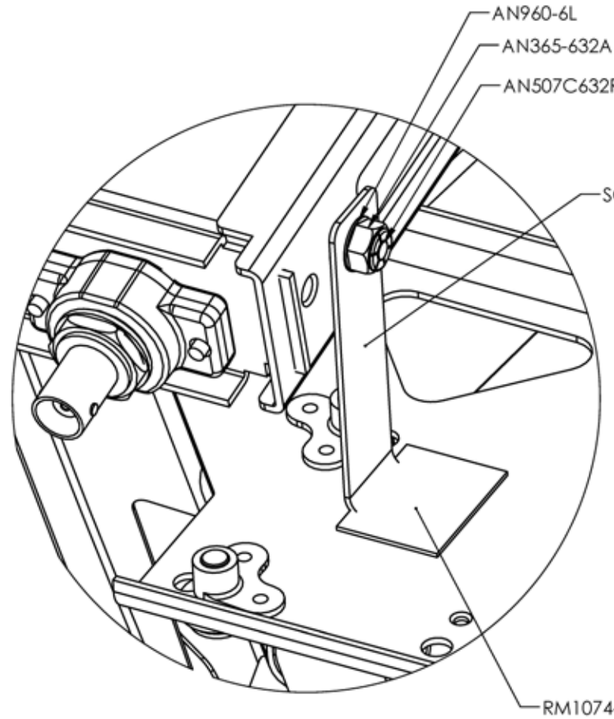








DETAIL H

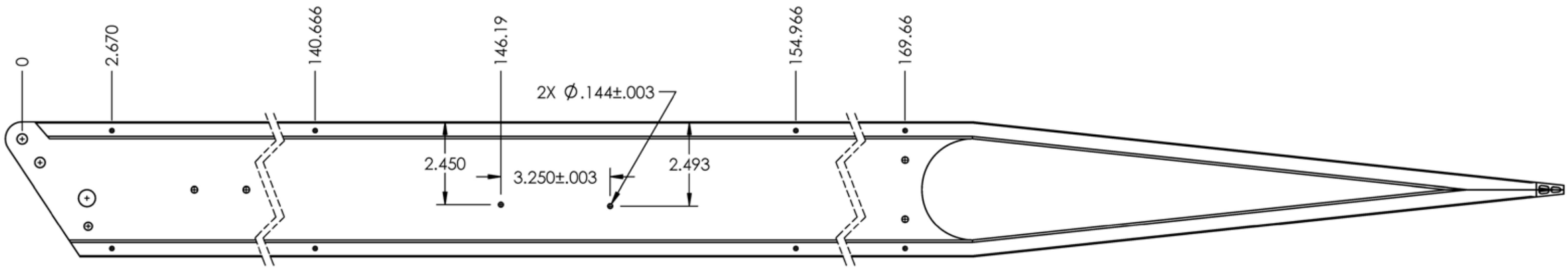


DETAIL J

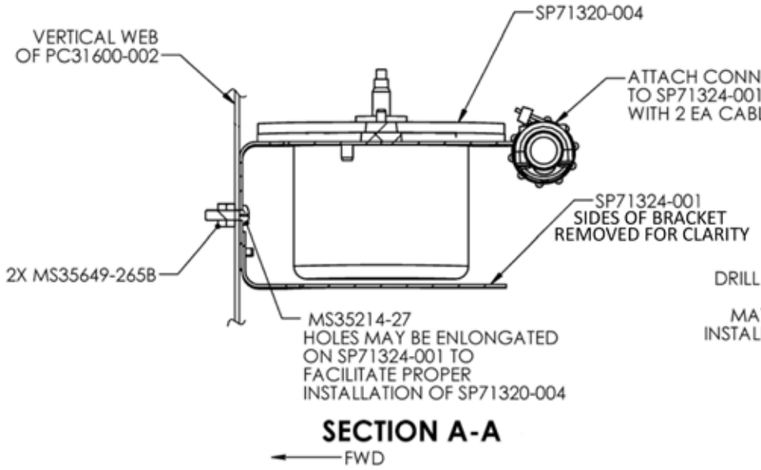
RADIO SUPPORT BRACKET INSTALLATION

EXECUTIVE GLASS TOUCH INSTRUMENT PANEL INSTALLATION

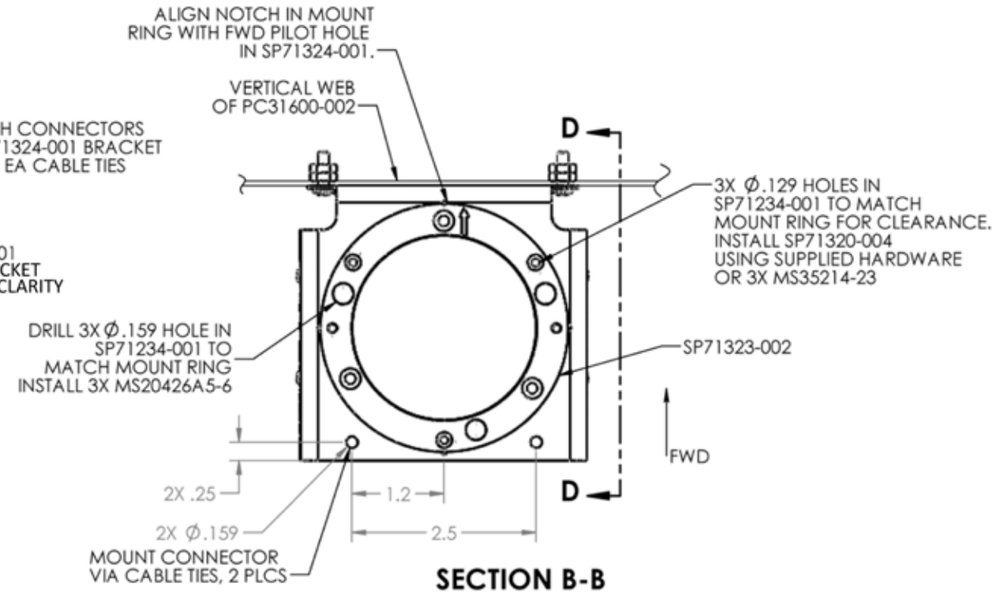
FWD FACE



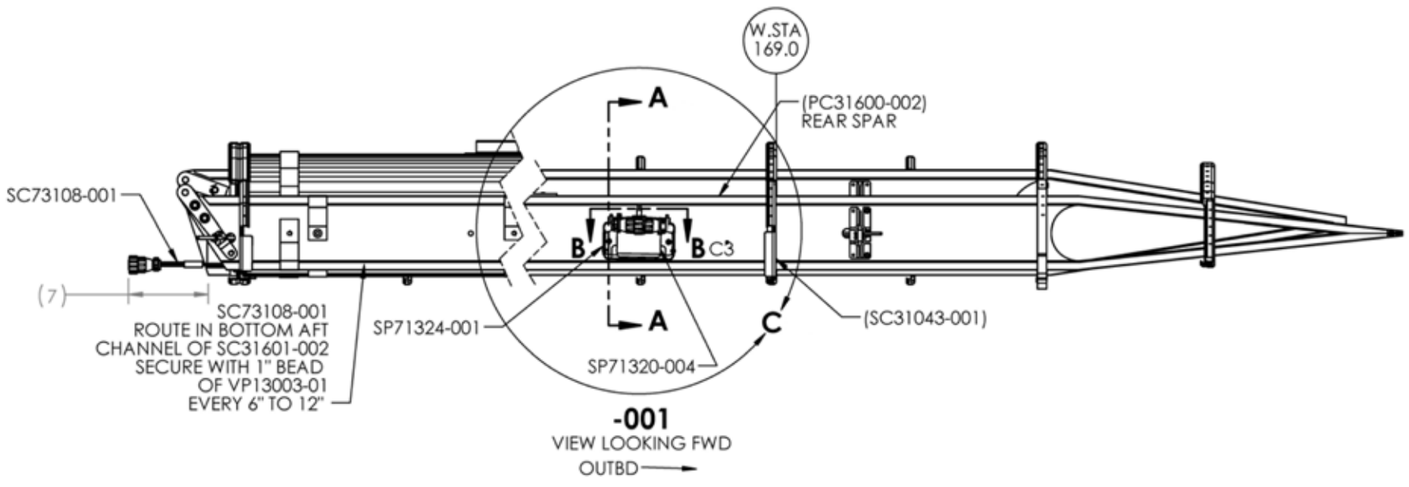
-002 RH ONLY
VIEW OF AFT FACE



SECTION A-A



SECTION B-B

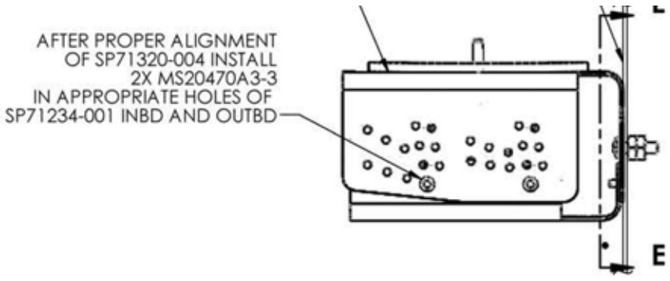


-001
VIEW LOOKING FWD
OUTBD →

NOTES:

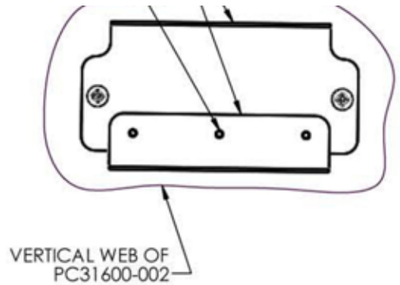
1. ADJUST THE SP1234-001 MOUNT TO BE LEVEL WITHIN 3.0° OF THE IN FLIGHT LEVEL CRUISE ATTITUDE.
2. ALIGN THE SP71234-001MOUNT'S FORWARD DIRECTION TO WITIN 0.5° OF THE LONGITUDINAL AXIS OF THE AIRCRAFT.

MAGNETOMETER INSTALLATION, RIGHT WING



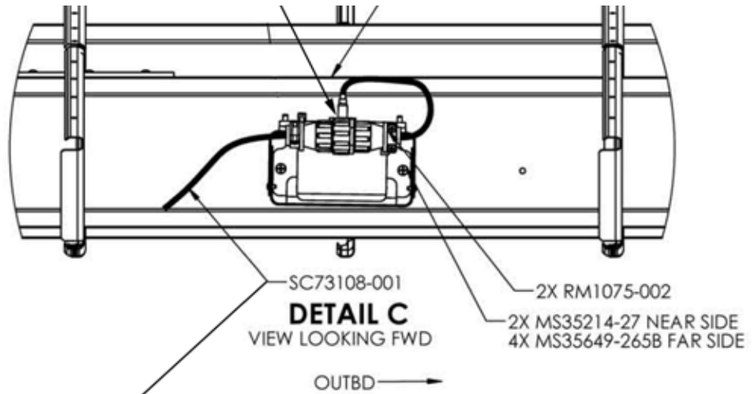
SECTION D-D

FWD →



SECTION E-E

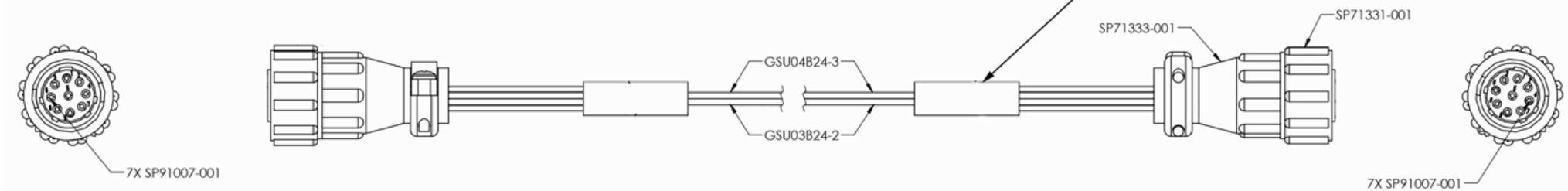
OUTBD →



DETAIL C

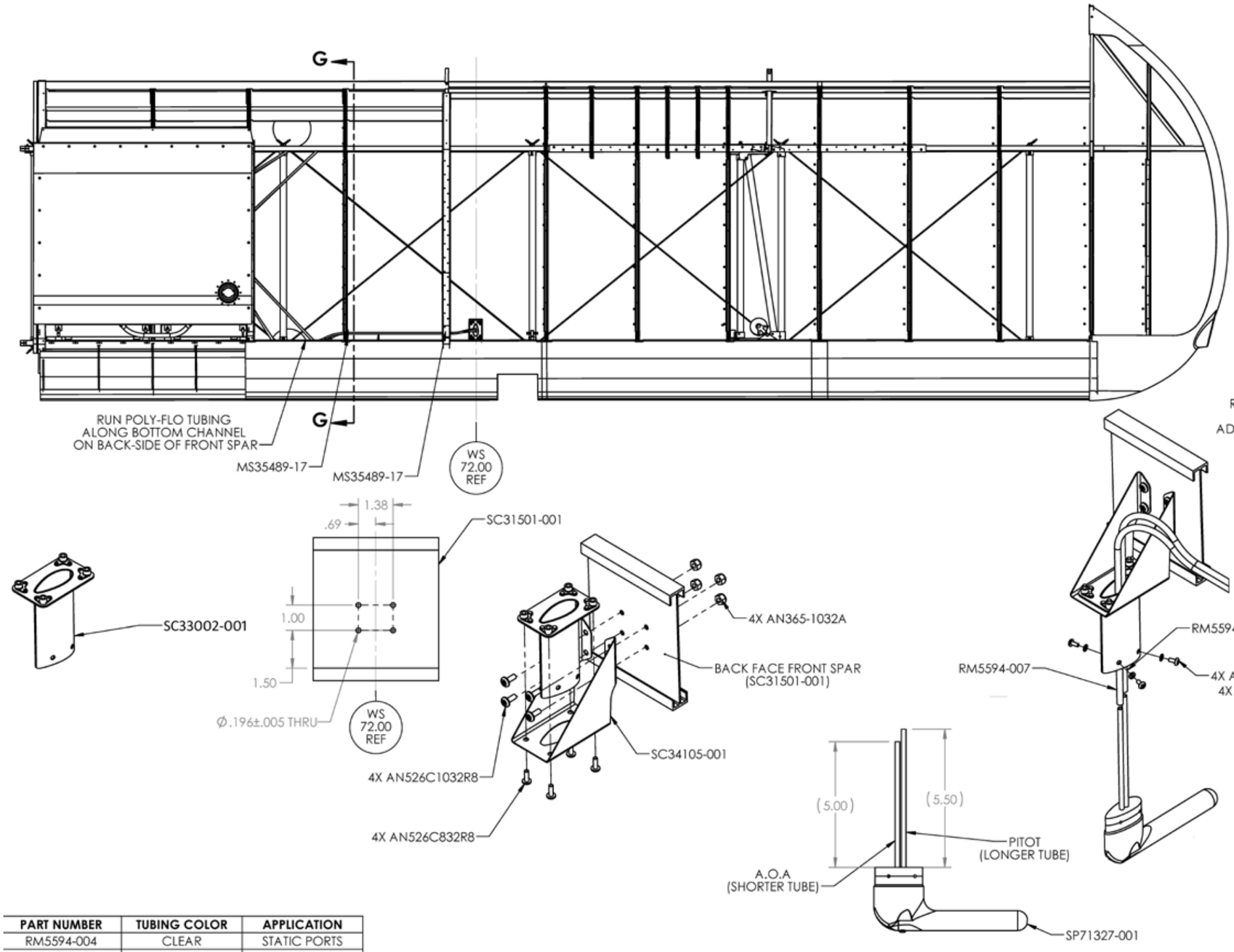
VIEW LOOKING FWD

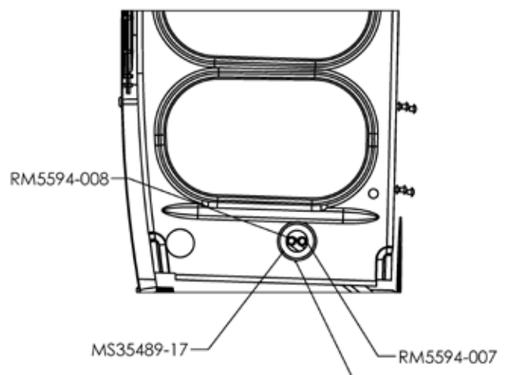
OUTBD →

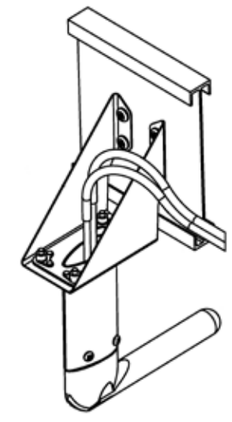


PIN #	WIRE COLOR	WIRE NUMBER	WIRE COLOR	PIN #
4	WHITE	GSU03B24-2	WHITE	4
2	WHITE/BLUE		WHITE/BLUE	2
1	JUMPER		JUMPER	1
8	WHITE/ORANGE	GSU04B24-3	WHITE/ORANGE	8
9	WHITE		WHITE	9
6	WHITE/BLUE		WHITE/BLUE	6
3	JUMPER		JUMPER	3

MAGNETOMETER INSTALLATION, RIGHT WING

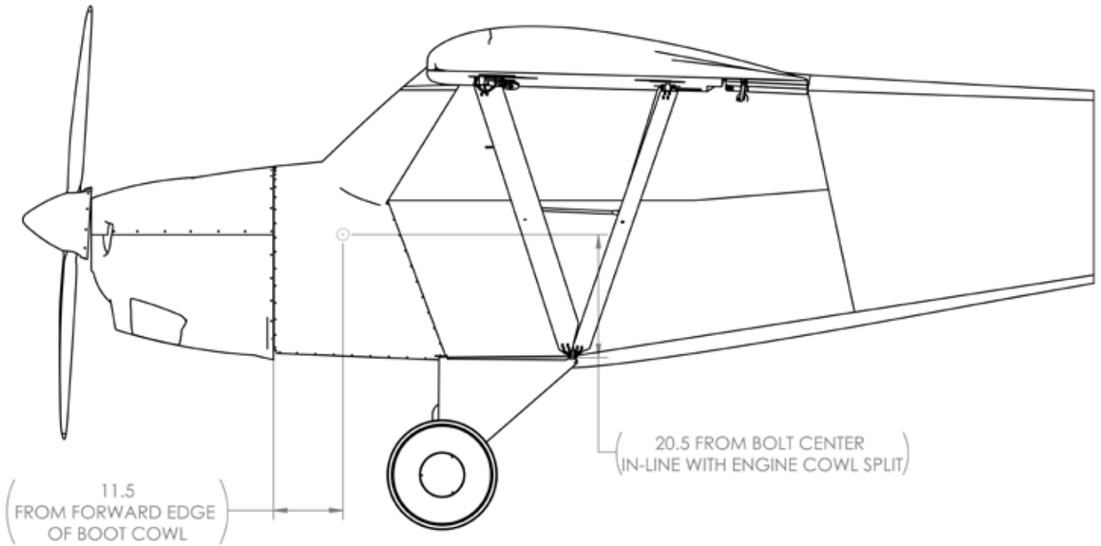


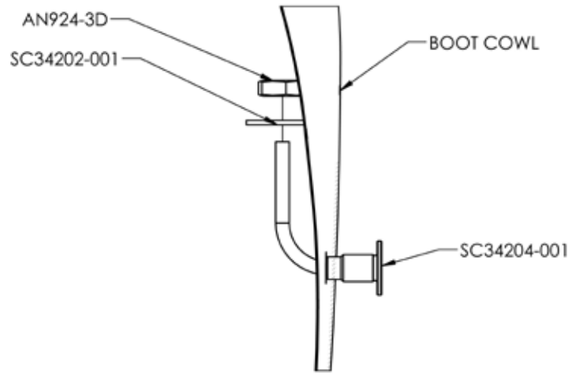


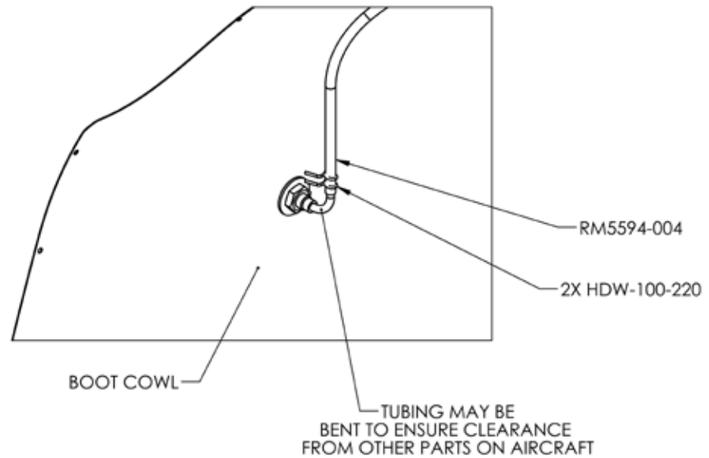


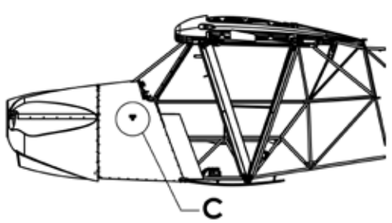
PART NUMBER	TUBING COLOR	APPLICATION
RM5594-004	CLEAR	STATIC PORTS
RM5594-007	GREEN	A.O.A
RM5594-008	BLUE	PITOT

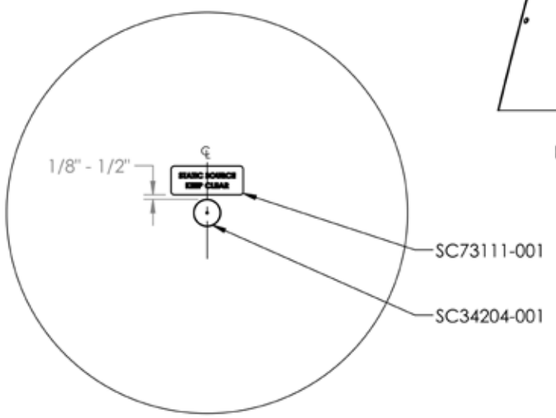
PITOT SYSTEM INSTALLATION









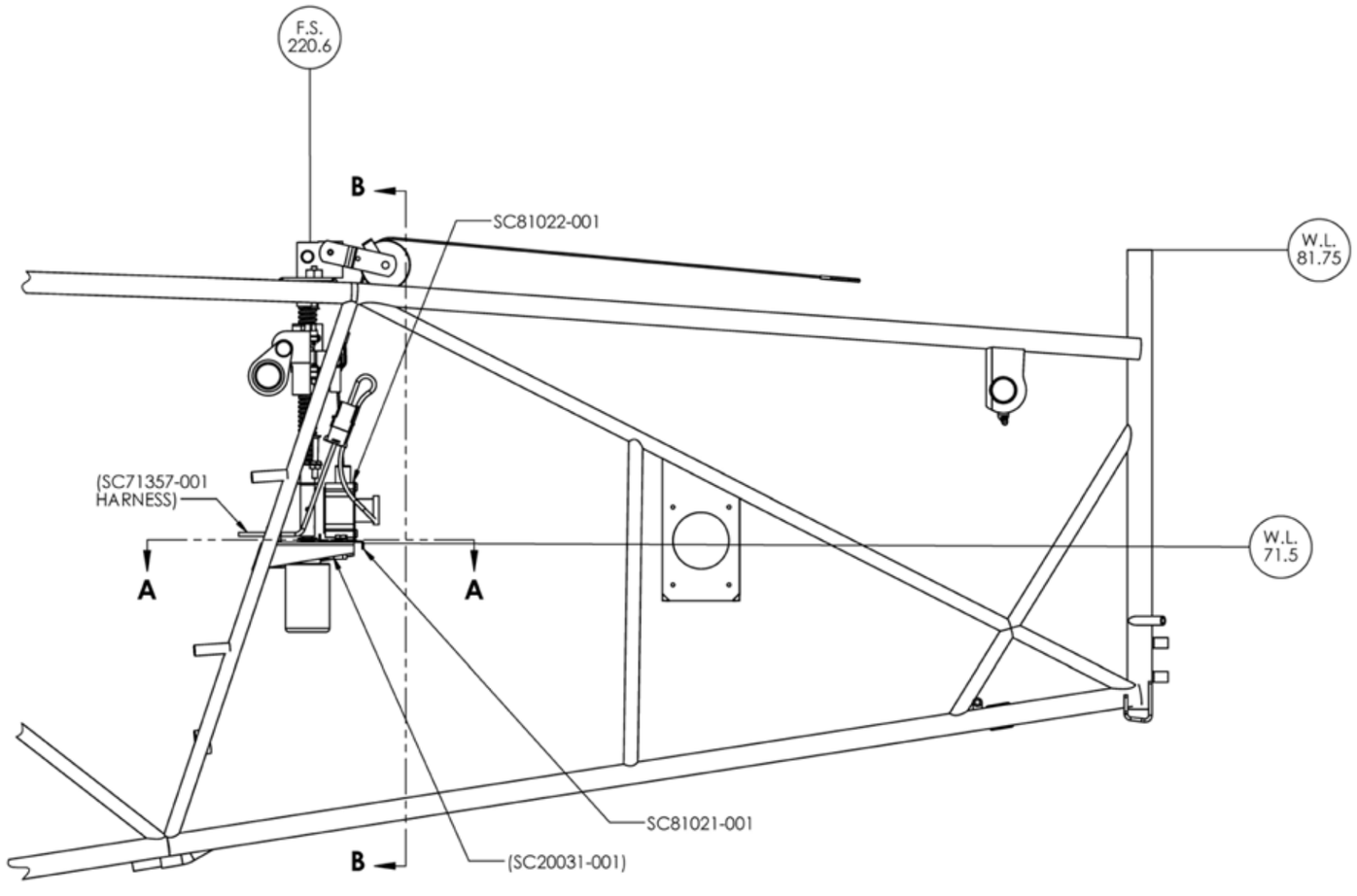


DETAIL C

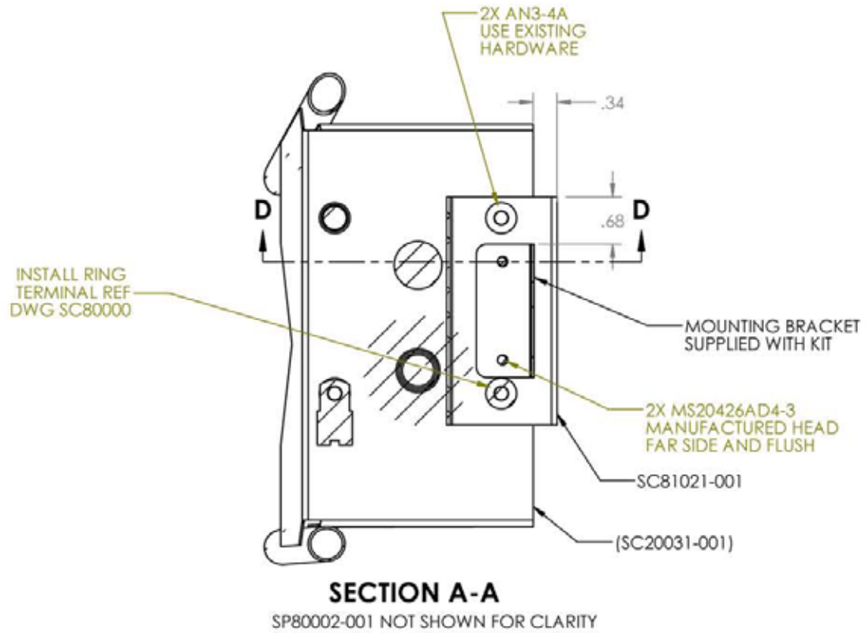
STATIC SYSTEM INSTALLATION

NOTES:

1. SOLDER WHITE/BLUE WIRE TO CCW TERMINAL, WHITE/ORANGE WIRE TO CW TERMINAL, WHITE WIRE TO RW TERMINAL.
2. USE HEAT SHRINK TUBING ON ALL SOLDER CONNECTIONS.
3. USE EXISTING HARDWARE, REPLACE AN3-5A WITH AN3-7A.

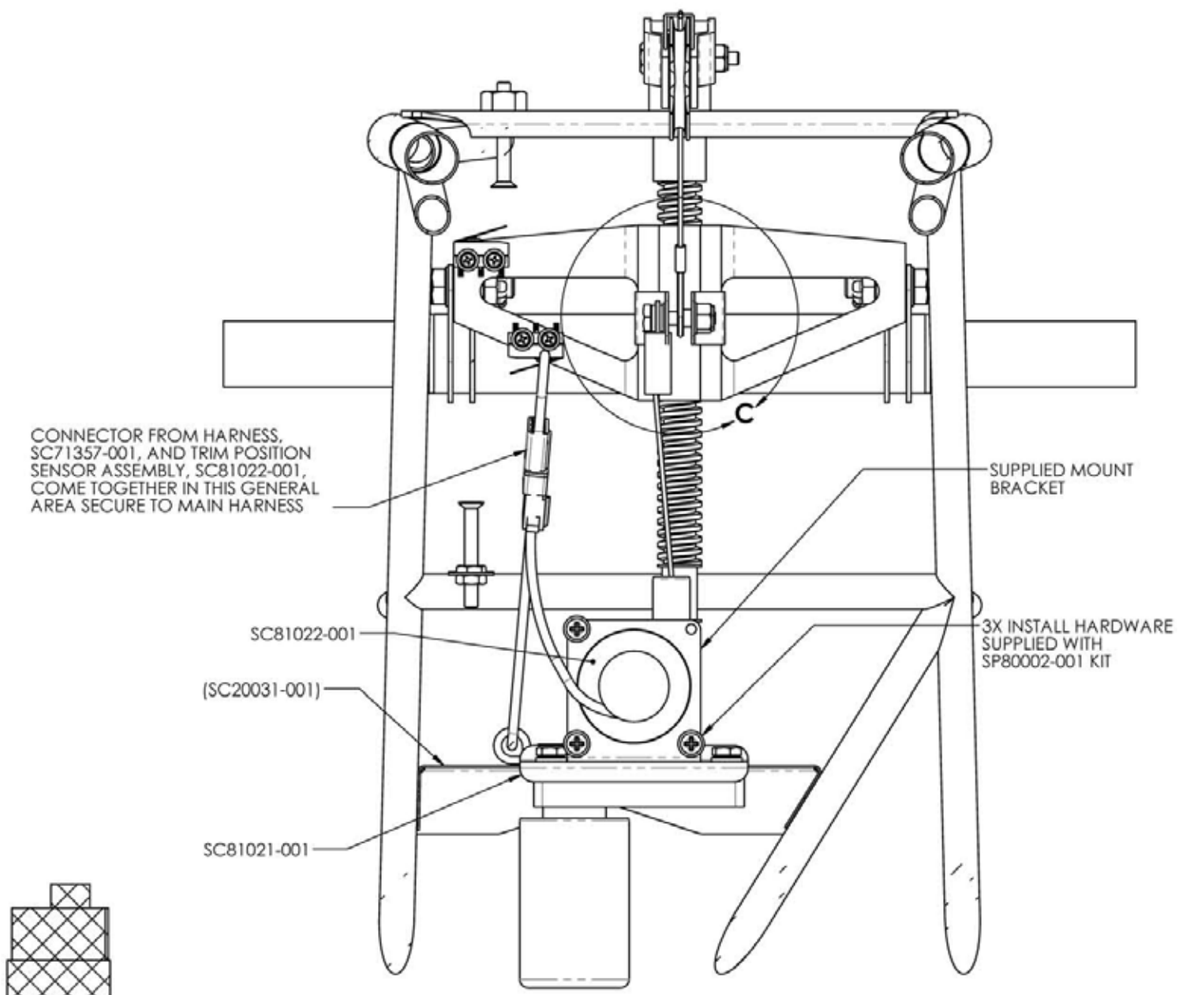


TRIM SENSOR INSTALLATION

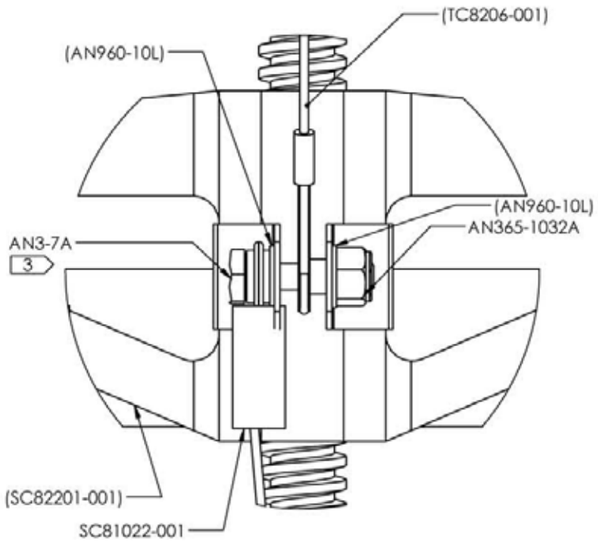


SECTION A-A

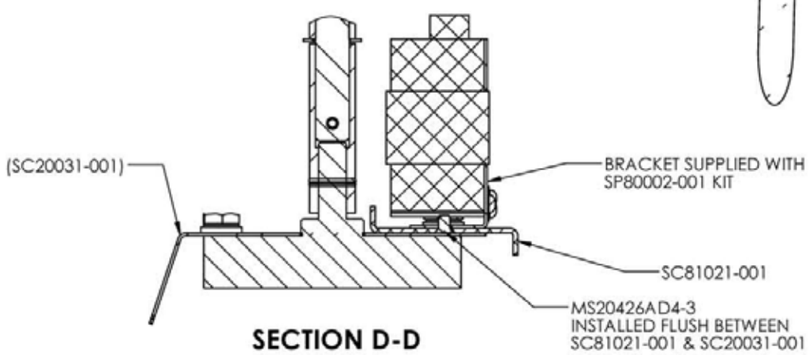
SP80002-001 NOT SHOWN FOR CLARITY



SECTION B-B



DETAIL C

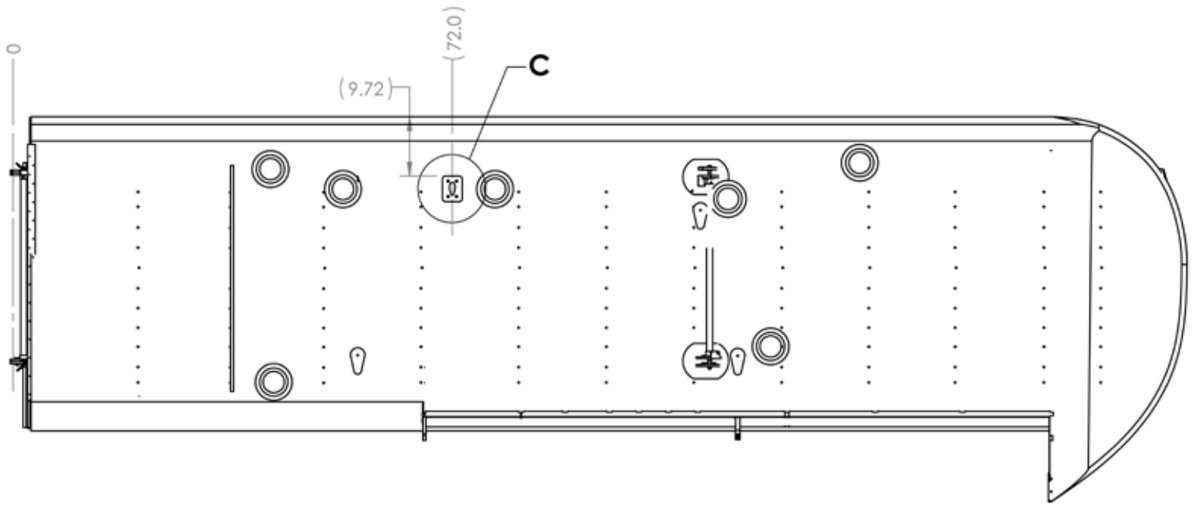


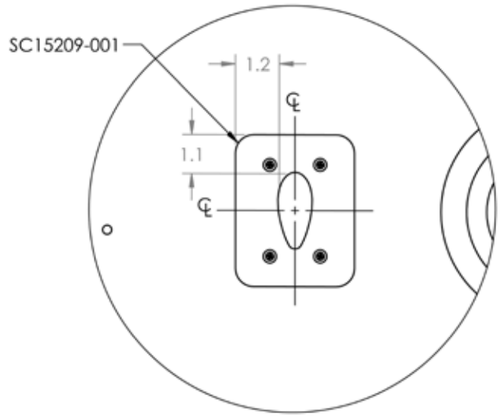
SECTION D-D

SP80002-001 NOT SHOWN FOR CLARITY

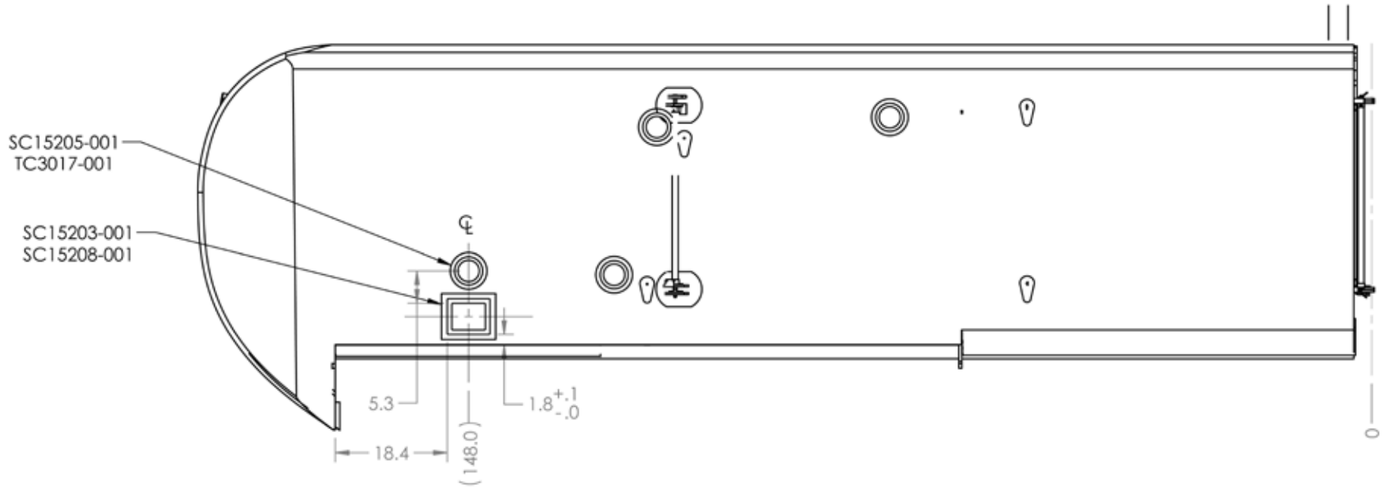
TRIM SENSOR INSTALLATION

ADDED PARTS FOR GARMIN EXECUTIVE GLASS TOUCH PANEL INSTALLATION ONLY



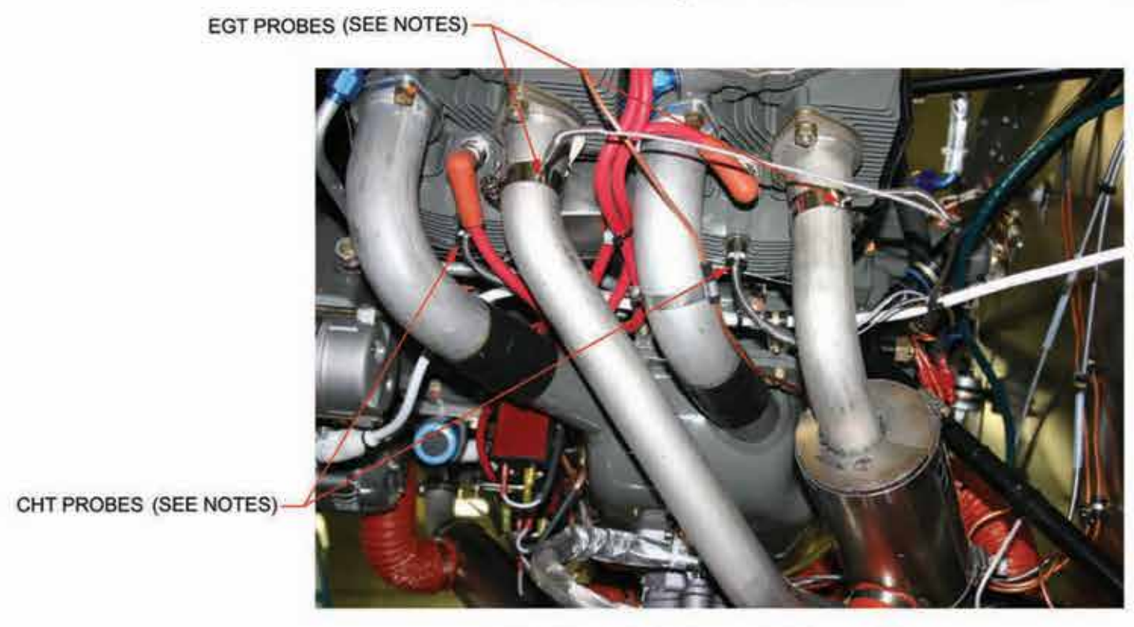


DETAIL C

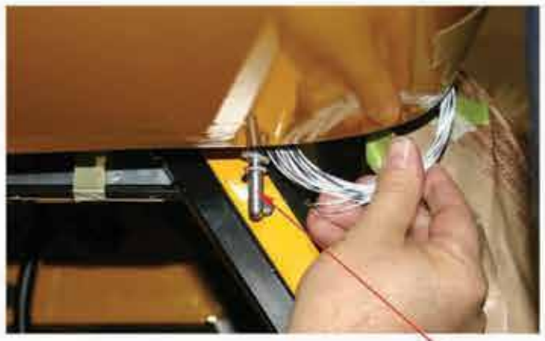


COVERED WING

RH CHT PROBE/EGT PROBE VIEW



LH CHT PROBE/EGT PROBE VIEW



OAT PROBE

DRILL 3/8" HOLE IN SKIN 2" AFT OF LEADING EDGE AND 1.5" OUT FROM WINDSHIELD ON THE RIGHT SIDE OF THE AIRCRAFT. STRING THE NYLON WASHER DOWN THE CABLE AND OVER THE OAT PROBE. THREAD ON THE NYLON NUT. CUT OFF THE ENDS OF THE OAT PROBE WIRES AND REPLACE WITH FEMALE DISCONNECT (VP6002-002.

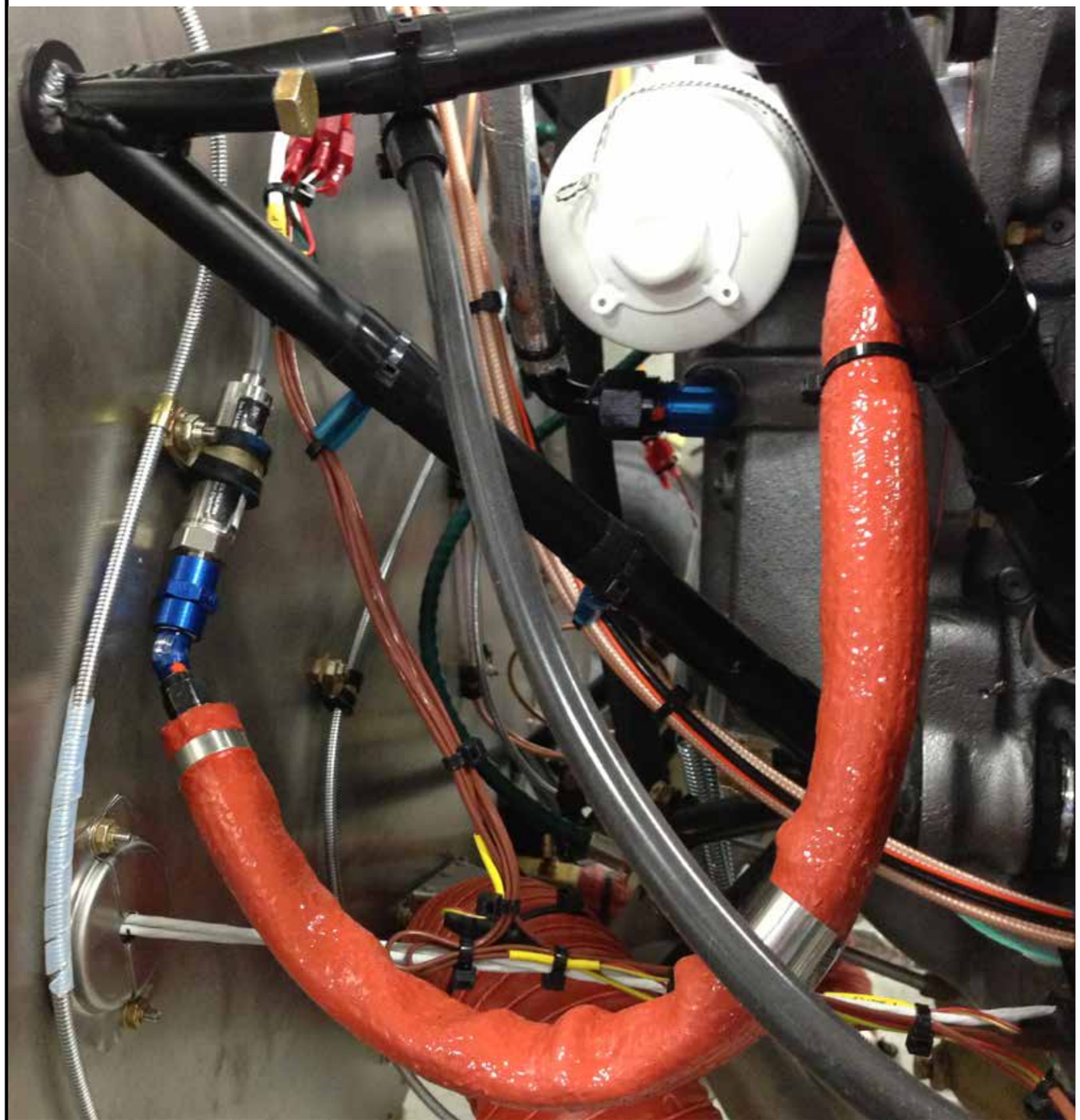
NOTES

ALL EGT PROBES SHOULD BE INSTALLED AT THE SAME RECOMMENDED DISTANCE OF 1.5" TO 2.0" FROM THE EXHAUST PORT.

ENSURE THE EGT PROBES WILL NOT INTERFERE WITH THE COWLING AND ACCOUNT FOR ENGINE SHAKE DURING STARTUP AND SHUTDOWN.

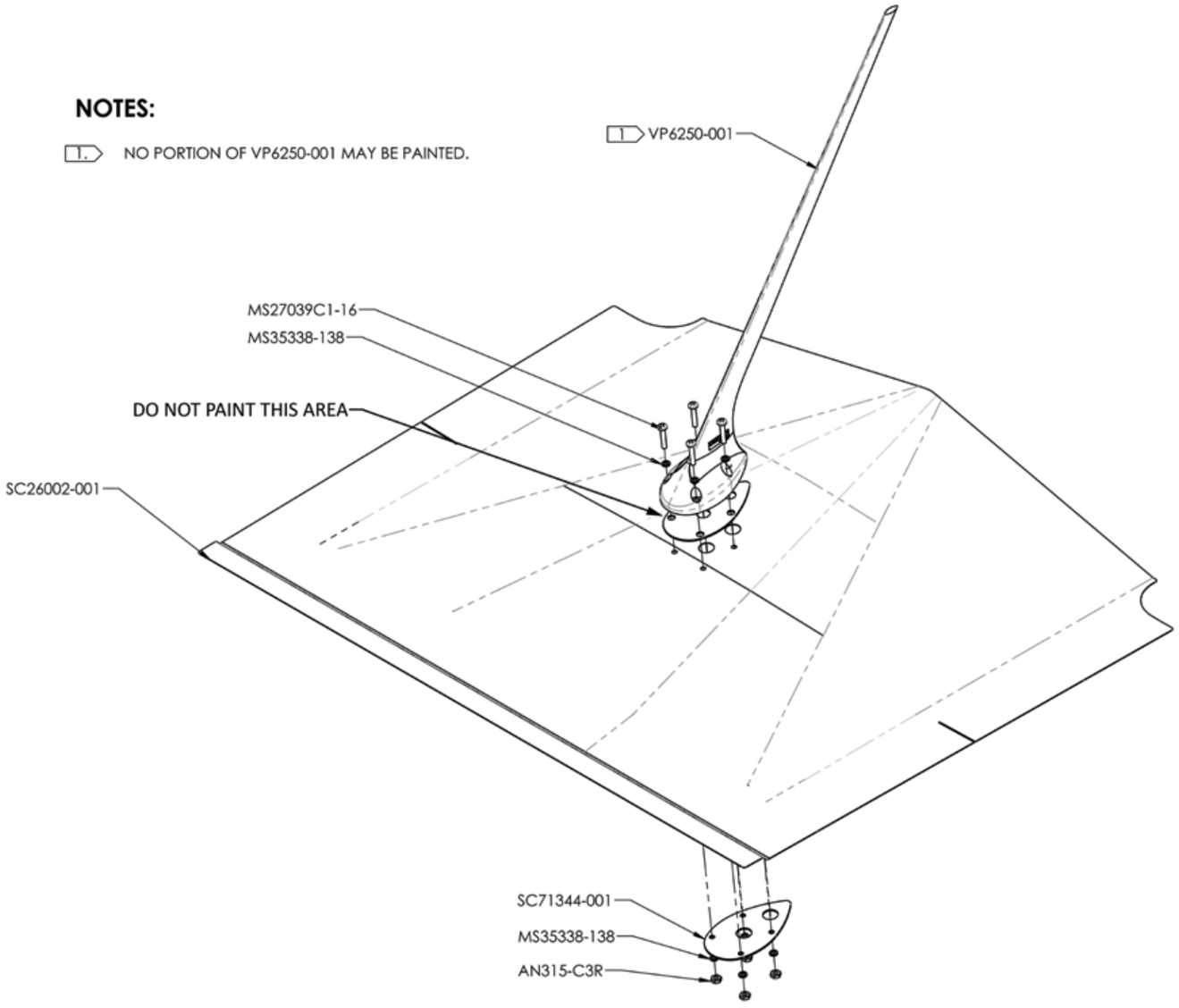
FOR EGT AND CHT PROBES, USE EXISTING DISCONNECT HARDWARE AND USE HEAT SHRINK TO COVER ANY EXPOSED METAL AS REQUIRED.

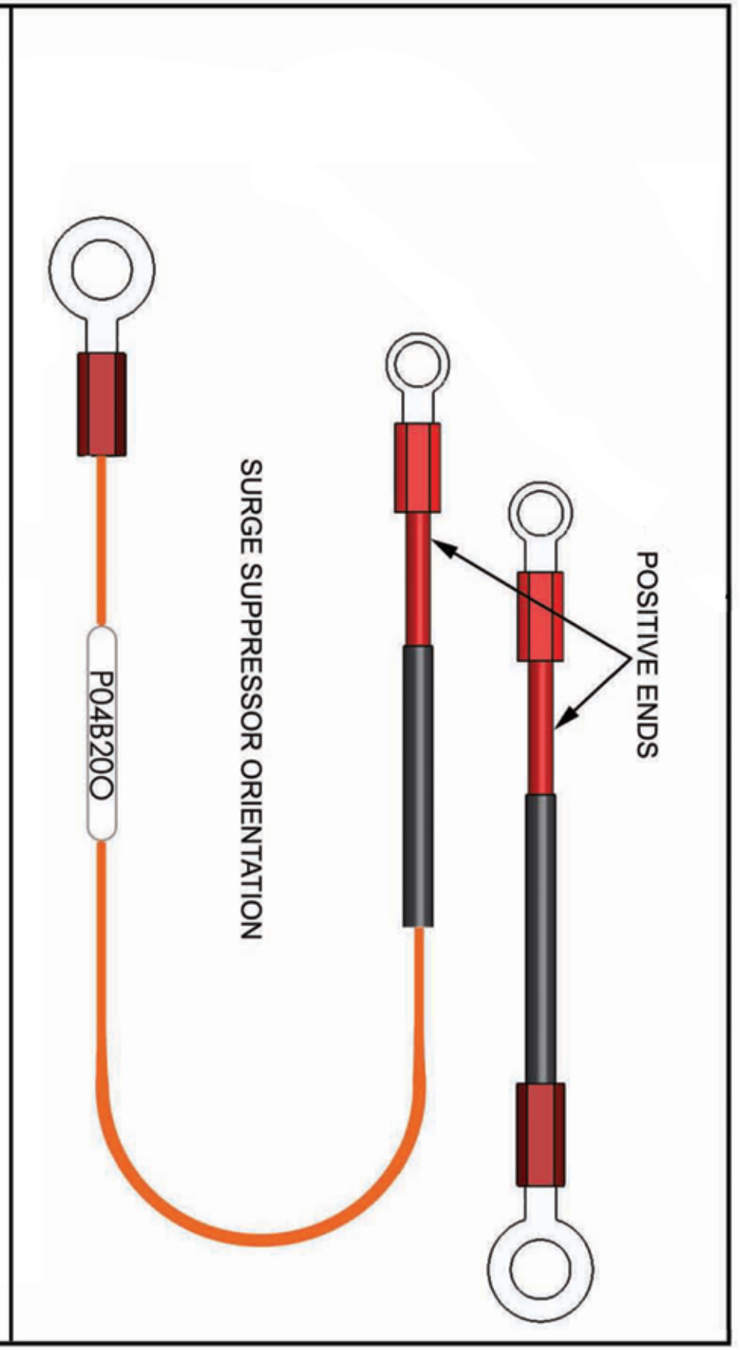
CRIMP MALE DISCONNECT CONNECTOR TO THE ATTACHING WIRES.

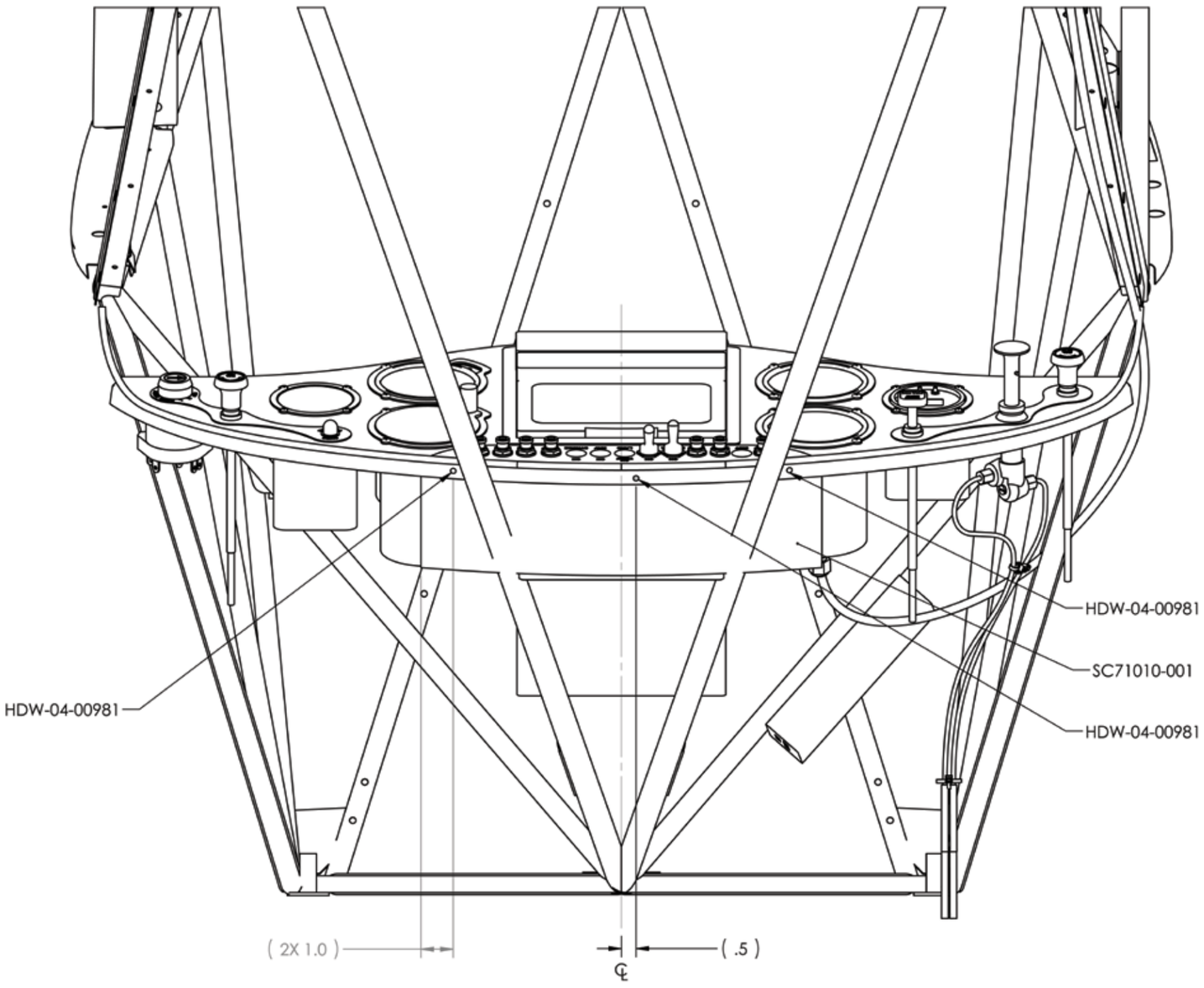


NOTES:

1. NO PORTION OF VP6250-001 MAY BE PAINTED.







ANTI-CHAFE STRIP INSTALLATION